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ASDIR-II VOLUME II PROGRAM DESCRIPTION

January 1975

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DEPUTY FOR DEVELOPMENT PLANNING
AERONAUTICAL SYSTEMS DIVISION
WRIGHT-PATTERSON AIR FORCE BASE, OHIO 45433

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This technical report has been reviewed and is approved for publication.

JAMES H. HALL Colonel, USAF

Deputy for Development Planning

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	programs used in ASDIR as well as a complete prog	
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TABLE OF CONTENTS

																														PAGE
TABLE	OF CON	TEN	(TS	·																										1
INTRO	DOUCTION													•																5
GENEI	RAL PROC	RAI	N D	ES	CR.	IP.	ΓΙ	OK									·													6
מסמס	RAM CONT	ו∩מי	ľ																											
I KOOL	ASDIR 2		• •	•	•	•	•	٠		•		•			•	•		•	•		•		•	•	•			•	•	7
DATA	INITIAL	ΙZ	AT I	CIN	A	ND	Ι,	/0																						
																														10
	DATINT																													11
	FILTER XPLOT																													12 13
	H20 .																													14
		:																												14
				. ~																										
HOT 1	PARTS AN SIGNIR																													15
	SIGNIK	•	• •	•	٠	•	•	٠	•	•	•	•	•	•	•	•	•	٠	•	•	•	•	•	•	•	•	•	•	•	10
PLUM	e gas dy																													
	PLMDEF																													27
	FLINP																													32
	CHEM	•		•	- •	•	•	•	٠	•	•	•	•	•	•	•	•	٠	•	•	٠	•	•	•	•	•	•	•	•	34 75
	THRUST PLMDM		• •																											35 37
	PLMPRN																													40
		•																												41
			- ~-										•																	
SIGN.	ATURE PI																													40
	ALPLUM RAYCAL																													42 44
	PLURAY																													51
	START																													53
	INTERP																													54
	PLUSIG							-			-				-										_	-	-	_	-	55
	ATMOS																										-	_	-	59
	SETTAU																			•										61
	TAUCAL																											,		62
																														63
	PLANCK																												-	64
	INTERP																													65
	TAUN20					•																							•	66
	ERF .	•	•	• •	•	•	•	•	•	•	•	•	•	•	٠	•	•	•	٠	•	•	•	٠	•	•	•	•	•	•	67
REFL	RENCES	•		•		•	•							•		•	•		•			•		•	•			•		68
APPE	NDIX -	PRC	GR.	AM	LI	ST	'IN	œ											•	•	•	•								A-1
OVER	LAY (XR		0	, (3)																									Δ-1

																										P	ACE
ARRAY		• • • • • • • • • • • • • • • • • • • •	•		 		•					 	 	•										•			A-7 A-15 A-32 A-36 A-30 A-39 A-40 A-41
OVERLAY (XFHP, INPUI	1,))			•			•																		A-43
RAYCAL - PLURAY - START - START - INTERP - PLUSIC - ATMS - SETTAU - LAUCAL - KECAL - PLANCK - INTERPO			• • • • • • • • • • • • • • • • • • • •		 			•		• • • • • • • • • • • • • • • • • • • •		 •	 	•									• • • • • • • • • • • • • • • • • • • •	• • • • • • • • • • • • • • • • • • • •	• • • • • • • • • • • • • • • • • • • •		A-47 A-50 A-54 A-57 A-61 A-63 A-69 A-70 A-71 A-73 A-75 A-76 A-77 A-78
OVERLAY (XRHP,	3, •	, (-	O) •				•					•				•				,							Λ-79
FLINP CHEM	•	•	•	•	•		•		•	:		•			:		•	•	•	•	:	•	•			•	A-81 A-87 A-91 A-93
OVERLAY (XRHP, PLMDM PLMPRNT . PLMPLT .	•	•	•								•																A-98 Λ-100 A-101
OVERLAY (XMP, FILTER .							•		•	•			•			٠.	•	•									A-100
OVERLAY (XRHP, XPLOI				•			•				•			•	•					•						•	A-109
OVERLAY (XRHP, REMAIN . AVER .					•	•				•						•	•		•				•	•		•	A-110 A-118

	PSOLN . STRATE . COFLOW . STOREI . STORE . AXAREA . GATHER . CPAGMA . PRNDTL .	• • •																				 					A-122 A-124 A-125 A-126 A-127 A-128
OVERI	LAY (XRHP, TRANCL . SETFLO .		•								:					:										:	A-130 A-133
OVERI	LAY (XRHP, FILMCL . SETFLO .		• ,																								
OVER	LAY (XRHP, CONFLM . PRNDTI .		•		•	•				:		•		•	•		:				•	•		•		•	A-139 A-147
	LAY (XRHP, CALFOR .	••	•	• ,			•				•			•			•	•	•	•	•	•	•	•	•		A- 1 48
OVER	LAY (XRHP, HOTPT . AVERHT . HEAT . TURBLT . HEATTR . AVERFS .		•	• 1	•	•	•		•	•																•	A-151 A-153 A-1 54 A-159
OVER	FINDIT . CODE CANSEE . BODY CHECK . RING . PICMAX . PICMIN .						• • • • • • • • • • • • • • • • • • • •		• • • • • • • • • • • • • • • • • • • •	• • • • • • • • • • • • • • • • • • • •		• • • • • • • • • • • • • • • • • • • •	• • • • • • • • • • • • • • • • • • • •	• • • • • • • • • • • • • • • • • • • •	• • • • • • • • • • • • • • • • • • • •	• • • • • • • • • • • • • • • • • • • •	• • • • • • • • • • • • • • • • • • • •									• • • • • • • • • • • • • • • • • • • •	A-176 A-177 A-180 A-181 A-182 A-183 A-184 A-186 A-188 A-189 A-190
OVER	LAY (XRHP, TASIB . CRODEL . MATPOS . MATSYM	• •	•	•				•										·									A-196

	WRIT14	•	•		•		•	•	•	•	•	•	•	•	•	•	•	:	•	•	•		•	•	•	•	A-199 A-201 A-202 A-203 A-204
OVER	LAY (XRHI	Ρ,	14	•	0)																						
	TAIL . PLOUT .																									-	A-205 A-296
OVER	LAY (XRHI TATLI . SIMPOW REDII . DIS RITE . VECANG		•			 				•										•	•	•	 •				A-208 A-209 A-210 A-215 A-216 A-217
OVER	LAY (XRHI TAILZ . PLANCK REDI . MINV . RITERI SUMATI				•	•	:	•	•		•		•	 		•		:	•				 •		•	•	A-218 A-219 A-220 A-222 A-224 A-225

ASDIR-II PROGRAM DESCRIPTION

INTRODUCTION

This Program Description outlines the structure of ASDIR-II in order to give the users the essential knowledge required for program implementation, use and modification. Simplified structural and flow diagrams of the major programs and short functional descriptions of all programs and subroutines are given to help describe ASDIR. The level of detail presented in this manual is consistent with the availability of reference and documentation material, program difficulty or program impact on overall computational accuracy.

Essentially this manual is written in the same order as the calling sequences in ASDIR, with modification only when structural considerations and clarity dictate.

Please note, this manual is intended as an aid to the user to allow better understanding of each program's function and not as a users manual (see ASDIR-II, Volume I - USERS MANUAL).

GENERAL PROGRAM DESCRIPTION

The Aeronautical Systems Division's InfraRed Signature Prediction Model (ASDIR) is a computer code utilizing the overlay structure shown in figure 1. These overlays are organized into three major functionally modularized computational groupings, figure 2: (1) Hot Parts Analysis; (2) Plume Gas Dynamics; and (3) Signature Prediction. A fourth overlay element establishes the required data initialization and input/output operational capabilities.

This volume is organized in the following functional format:

- 1. ASDIR 2 Program Control
- 2. Data Initialization & I/O
- 3. Hot Parts Analysis
- 4. Plume Gas Dynamics
- 5. Signature Prediction
- 6. Program Listing in Overlay Form

ASDIR 2 (Page A-4)

FUNCTION:

This program is the main controlling routine of ASDIR (the Aeronautical Systems Division InfraRed Signature Prediction model). ASDIR's prime function is to provide a logical sequence of calls to the functional programs.

INPUT:

None

OUTPUT:

None

SUBROUTINES:

All overlays and data blocks

DESCRIPTION:

See figures 1 and 2 and program listing on

page A-4.

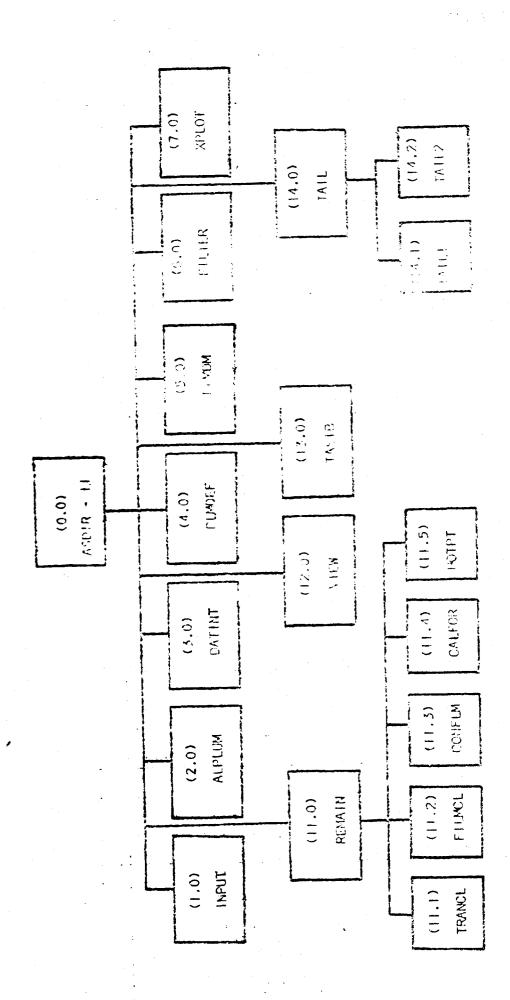


FIGURE 1. ASDIR-11 OVERLAY STRUCTURE

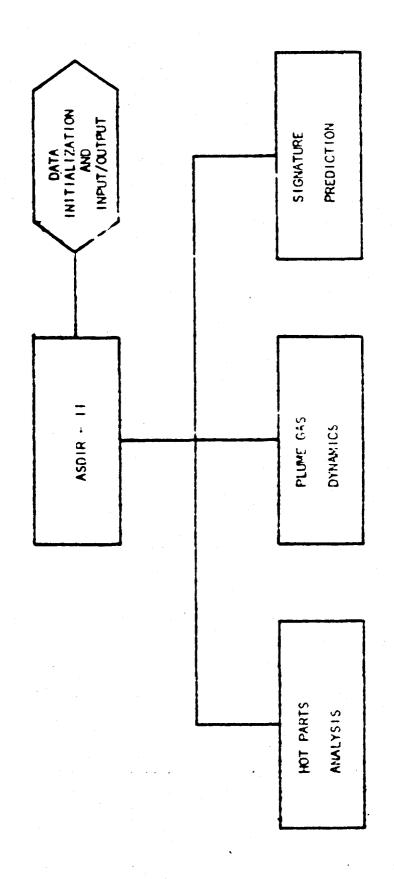


FIGURE 2. ASDIR-11 FUNCTIONAL STRUCTURE

INPUT (Page A-43)

FUNCTION:

Provide a vehicle by which the desired calculation parameters, case definition and data initialization

can be read into ASDIR.

INPUT:

All control and computational variables in the

CASE namelist.

OUTPUT:

Computationally compatible case and dimensional

data.

SUBROUTINES:

None

DESCRIPTION:

See ASDIR-II Volume I, User's Guide, for complete

description of the namelist variables.

DATINT (Page A-79)

FUNCTION:

To initialize the calculation parameters.

INPUT:

None

OUTPUT:

Initialized calculation parameters

SUBROUTINES:

None

DESCRIPTION:

This program has been written to initialize the common block data to the default case. See ASDIR-II, Volume I, User's Guide for default options. A secondary function of this program is to allow use of ASDIR on systems in which a zero core initialization is not available.

FILTER (Page A-106)

FUNCTION:

Provides a means of accessing stored filter response data or enturing a filter response

of the user's choice.

INPUT:

Filter number or filter band width and response.

OUTPUT:

Filter response.

SUBROUTINES:

) ie

DESCRIPTION:

Stored filter responses are in data statements with modification possible through namelist FILT. This program also can be used to initialize the wavenumber calculation intervals.

XPLOT (Page A-109)

FUNCTION:

To provide a means of file manipulation for either punching or printing spatial signature

data.

INPUT:

Calculated spatial signature data.

OUTPUT:

Punched or printed spatial data.

SUBROUTINES:

None

DESCRIPTION:

This program requires the use of a punch file

and an addition file TAPE 8.

Block Data H20 (Page A-15)

FUNCTION:

Data Block for water vapor band parameters.

INPUT:

None

CUTPUT:

None

SUBROUTINES:

None

DESCRIPTION:

See References [1, 2, 3]

PROGRAM:

Block Data CO2 (Page A-7)

FUNCTION:

Data Block for carbon dioxide band parameters.

INPUT:

None

OUTPUT:

None

SUBROUTINES:

None

DESCRIPTION:

See References [1, 2, 3

SIGNIR

FUNCTION:

To calculate the hot parts signature and descriptive parameters required in ASDIR.

INPUT:

Engine description, gas properties and case definition (See ASDIR-II, Volume I, User's Manual).

OUTPUT:

Primarily the information which is transferred from the SIGNIR link to ASDIR is an equivalent black body area and equivalent black body temperature.

SUBROUTINES:

See Figure 3.

DESCRIPTION:

With the exception of the modifications required to adapt SIGNIR to the ASDIR format the hot parts calculations are performed with the original LTV/SIGNIR-I computer code. This code was chosen for only one reason; it has proven to be the best available methodology to predict engine hot metals signatures regardless of program size, operating cost or complexity.

The description of the infrared signature prediction program (SIGNIR) is provided by the following discussions which cover a general description of the program, the program computational procedure and the analytical methods utilized in obtaining the solution.

The program SIGNIR is a digital computer program written for the purpose of providing predictions of the hot metal infrared emission from aircraft engine exhaust systems. The program is designed to be applicable to axi-symmetric turbojet, turbofan, or turboshaft engine exhaust systems. It predicts the spectral intensity of the radiant energy emitted from exhaust system hot metal in the wavelength band of from one to 15 microns. In general, the information required by the program is as follows:

- . Exhaust system physical characteristics
- . Engine operating conditions
- . Special surface cooling flow conditions
- . Exhaust system surface properties

The predictions provided by the program for the combination of a selected maximum of 20 engine off-axis angles are:

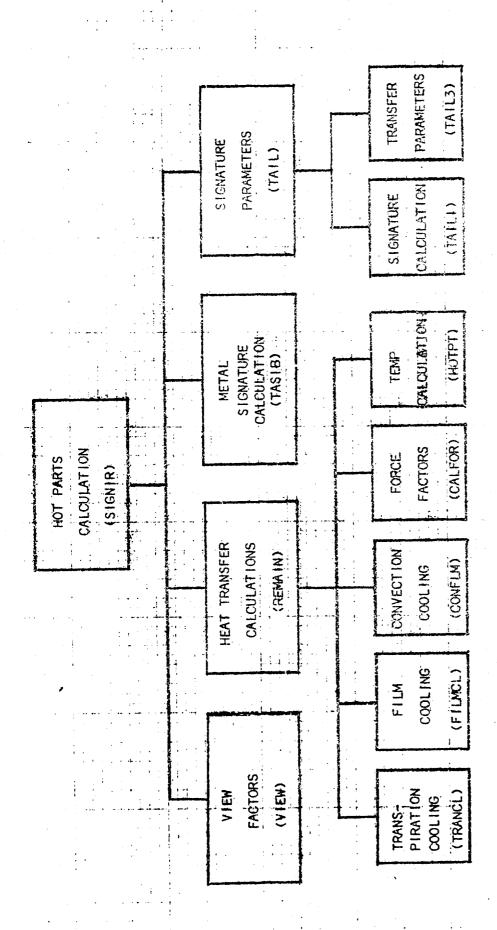


FIGURE 3. HOT PARTS ANALYSES

Spectral radiant intensity nonattenuated by the atmosphere Selected radiant intensity bandwidth summations nonattenuated by the atmosphere

Optional exhaust system information which can be requested from the program is:

. Internal fluid flow properties

. Surface boundary layer data

. Internal and external geometric radiation view factors

. Surface temperature distribution

. Thrust information

The process of computing the intensity of the infrared energy emitted from aircraft exhaust system hot metal requires a knowledge of the metallic surfaces temperature, emissive properties and their relative physical positioning within the system. From this information, the radiant energy emitted and reflected from the exhaust system to a point in space can be determined. The computational procedure established by this program for the prediction of exhaust system hot metal infrared radiation is shown by major subjects in the general block diagram presented on figure 3.

The input data required by the program provides a physical model (nodal system) of the exhaust system configuration, the properties and conditions of the exhaust gases entering the system, band width and location relative to the exhaust system. The program undertakes a one-dimensional, isentropic, compressible flow analysis which generates the axial distribution of exhaust gas flow conditions and properties within the system. Where two gas streams at different temperatures flow within a single passage, compound flow considerations are utilized. The program adjusts the flow at the nozzle throat to balance with the ambient pressure or a chocked condition dependent on the system's entrance flow data.

With the internal flow properties defined, a surface boundary layer analysis is conducted to obtain the coefficients of convection and friction along each surface within the system. This analysis is primarily dependent on the surface's axial static pressure distribution and the upstream boundary layer conditions. An area weighted average coefficient of convection for each of the model's surface nodes (isothermal regions established by the input data) is computed for the system thermal analysis. The axial force on each surface, internal to the system, is computed utilizing both the frictional and pressure-area forces.

Descriptions of the routines associated with SICNIR are presented in Table 1. Further definitions of the routines can be found in ASDIR-II, Volume III, Reference Documentation or in Reference [4].

The program SIGNIR, other than a few minor programming changes, remains as developed by LTV. The only link between SIGNIR and ASDIR-II is Common Block OUT, containing the following information:

. Equivalent Black Body Area, CM²

. Equivalent Black Body Temperature, deg-K

Viewing Aspect Angles, degrees off tail

. Number of Aspect Angles

TABLE 1 DESCRIPTION OF SIGNIP ROUTINES

ROUTINE NAME	BASIC FUNCTION	PAGE
ARCCOS	Determines the angle, in radians, corresponding to a given cosine value. The angle will be from zero to H/2 radians.	A-176
ARRAY	Describes in two-dimensional coordinates the inner and outer surfaces of the flow streams using the surface node coordinates.	A-38
AVFR	Averages the cooling wall temperatures for the surface nodes along the cooled portion of the configuration surfaces; prints these averages.	A-118
AVERFS	Determines the average values of the parameters necessary for force factor (pressure area and friction forces) calculations for each surface node along the configuration surfaces.	A-160
AVERHT	Determines and prints the average heat transfer coefficient for each surface node along the configuration surfaces.	A-151
AXAREA	Determines the cross-sectional area of the fluid flow streams as a function of the configuration axial distance.	A-126
BODY	Determines for the internal view factors, if the line of sight between points on two nodes is tangent to a shadowing body.	A-186
CALFOR	Prints the average force factor for each surface node and calculates and prints the total net surface force factor for the	A 140
CANSEE	configuration. Checks, for internal view factors, the visibility between two nodes to determine if shadowing may occur.	A-148 A-184
CEDIT	Selects the convection and conduction heat paths from input data and forms the conduction matrix.	A-203
CHECK	Checks the data returned from routine BODY for extraneous solutions.	A-188

ROUTINE NAME	BASIC FUNCTION	PAGE
CODE	Analyzes the results provided by routine FINDIT to describe the nature of shadowing between two nodes.	A-183
COFLOW	Calculates the compressible flow properties for the flow streams as a function of pressure. Adjusts the flow for cooling or ambient pressure at the exit of the configuration.	A~122
COND	Reads in those values of heat transfer and fluid lump temperatures for nodes whose thermal paths are not connected with the fluid streams.	A-4 0
CONFLM	Calculates a pressure balance and surface temperatures for a surface cooled by convection-film cooling for either counter or parallel flow. Prints the coolant mass flow rate and pressure.	A-139
CPAGMA	Calculates the specific heat at constant pressure as a function of temperature for routine HEAT and the specific heat ratio as a function of Mach number of routines RIMAIN and HEAT.	A-128
CRODEL	Computes Kroneker Delta for matrix solutions,	A-196
CUBIC	Determines the real roots of a third order equation.	A-178
DINVRT	Computes the matrix inversion and accompanying solution for linear equations.	A-199
DIS	Computes the distance between two given coordinates.	A-215
FEDIT	Selects the radiation view factor data from the input data and forms the radiation matrix.	A-202
FILMCL	Calculates a pressure balance and surface temperatures for a film cooled surface. Prints the coolant mass flow rate and pressure.	A-134

ROUTINE NAME	RASIC FUNCTION	PAGE
FINDIT	Locates the axial positions of two nodes with respect to the axial positions of the configurations surfaces.	A-182
FLOWNO	Checks the resulting pressure for surface cooling against the static pressure along the cooled surface. Prints a warning diagnostic for smeas of no cooling flow.	V-118
GATHER	Combines data from routines AXAREA and COFLOW to form the compressible flow properties for one configuration flow streams.	A-127
HEAT	Calculates the initial conditions necessary for the boundary layer calculations.	A-153
HEATTR	Calculates the convection heat transfer coefficient.	A-1 59
нотрт	Routine to control the calculation of surface boundary layers and heat transfer coefficients.	A-1 4 9
MATPOS	Checks to ensure that all the elements for convection and radiation matrices are positive.	A-197
MATSYM	Completes the matrices for the convection heat paths and the radiation view factor area product.	A-198
MINV	Inverts the "Script F" matrix.	A-222
ORDER	Arranges two-dimensional coordinate arrays into increasing order according to a selected variable.	A-39
PICMAX	Determines the maximum value of an array of up to twenty numbers.	A-1 90
PICMIN	Determines the minimum value of an array of up to twenty numbers.	A-191
PLANCK	PLANCK Black body function.	A-219
PLOT1	Generates a plot of radiant intensity vs. wavelength for both the non attenuated and the atmospheric attenuated radiation data.	A-206

ROUTINE NAME	BASIC FUNCTION	PAGE
PRNDT1	Calculate the Prandtl number as a function of termperature for routine CONFLM.	A-147
PRNDTL	Calculates the Pradtl number as a function of temperature for routines TURBLT and RIMAIN.	A-129
PSOLN	Generates surface cooling flow supply and discharge characteristics and solves these simultaneously for calculation of the available surface cooling mass flow rate and pressure.	A-120
QUAD	Determines the roots of a second order equation.	A-177
REDI	Computes the spectral radiant intensity (watts/Steradian) for each engine off-axis angle.	A-220
REDII	Computes and prints geometric view factors for a detector to the configurations surface nodes as a function of engine off-axis angle.	A-210
REMAIN	Control routine for the fluid flow, heat transfer, surface cooling and surface force factor calculations. Prints the fluid flow information.	A-110
RING	Determines the intersection of a shadowing body with the "cone of vision" that a point on one node secs another node.	A-1 89
RITE	Used to print the heading information for routine REDII.	A-216
RITER1	Prints the surface radiation information.	A-224
SCRPTF	Computes internal radiation interchange factors for the thermal analyzer.	A-204
SETFLO	Determines the pressure balance for the surface cooling routines if a cooling mass flow rate is specified.	A-133, A-138
SHADOW	Describes the configuration surfaces used for shadowing in routine VIEW in a manner compatible for shadowing surface description of routine REDII.	A-209

ROUTINE NAME	BASIC FUNCTION	PAGE
SIGNIR	The programs main routine. Controls the flow throughout the entire program.	A-32
STORE	Temporarily stores information necessary for average heat fransfor coefficient calculations.	A-125
STOREL	Temporarily stores information recessary for force factor calculations.	Λ-124
STRATE	Computes solution utilizing linear interpolation routine used in describing the configuration surface data for boundary layer calculations.	۸-121
SUMAT1	Summates and prints the adiant intensity for selected wavelength bands.	A-225
TABLEL	Computes solution utilizing second order interpolation of data and provides the first derivative of the interpolated function.	A-36
TAIL	Interface parameter generation and I/O program control routing.	۸-205
TAIL1	Metal signature calculation control routing.	A-208
TAI L2	I/O and interface parameter calculation control routine.	A-218
TAS1,R	Controls program flow and data involved in calculating the radiation interchange factor matrix, conduction matrix, performing the thermal balance and printing the wall temperature for each side.	A-19 3
TEST1	Tests the possibility of one node being shadowed by a second node which has the same axial location.	A-180
TEST2	Uctermines whether a point lies between two other points.	Λ-181
TESTN	Computes the last computed value of geometric view factor with the previously computed value to determine whether the view factor needs to be recalculated using finer divisions of the nodes. This routine also sets these divisions	A_103

ROUTINE NAME	BASIC FUNCTION	PAGE
TITLA	Controls output printing of the title page.	A-41
TRANCL	Calculates a pressure balance and wall temp- eratures for a transpiration cooled surface. Prints the coolant mass flow rate and pressure.	A-130
TURBLT	Calculates and prints the boundary layer and heat transfer information as a function of configuration axial distance.	A-154
VECANG	Calculates the angle of a node relative to the configuration center line.	A-217
VIEW	Calculates and print: for each configuration node, internal geometric veiw factors.	A-162
WRIT14	Transfers a two-dimensional array into a single dimensional array.	A-201

The equivalent Black Body Area and Temperature are extracted from the data generated by SIGNIR. The first step is to determine the equivalent Black Body Temperature. This is accomplished by searching the generated spectral data for the maximum intensity, figure 4, and applying the Wien Displacement Law:

$$T_{BR} = \frac{2897.9}{max} v^{-9} X$$

The equivalent Black Body Area is found by dividing the integrated signature obtained from SIGNIR by the Radiant Exitance over the band of interest.

$$A_{BB} = \int_{\lambda_1}^{\lambda_2} M_e(SIGNIR) / \int_{\lambda_1}^{\lambda_2} M_e(T_{BE})$$

Figure 5 shows a typical correlation between the predicted spectral signature and the spectral signature of the equivalent black body. These resultant temperatures and areas are the only parameters transferred back to ASDIR.

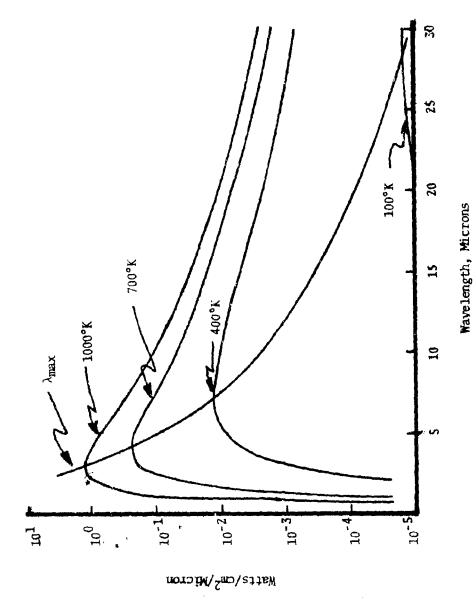
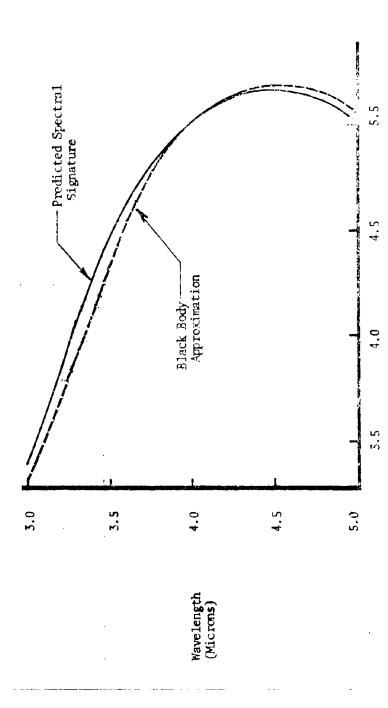


Figure 4. Black Body Curve



Watts/Steradian Figure 5. Black Body Correlation

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PLMDEF (Page A-81)

FUNCTION:

Plume definition and gas data array development.

INPUT:

Engine nozzle exit flow properties (velocity, pressure, temperature, and species [H20, CO2] concentrations) as

a function of radius.

OUTPUT:

Plume gas data array. At this point, velocity (ft/sec) and pressure (atm x 10^{-3}) are multiplexed into common

registers.

SUBROUTINES:

FLINP

DESCRIPTION:

The gas dynamic calculations made for the plume in this program, figure 6, are applicable to axisymmetric turbojet and turbofan engine with mixed or separate co-annular co-plonar exhaust jets. The finite difference calculations were developed to express the conservation (mass, momentum, energy, and species) laws for numerical integration by General Electric in their early SCORPIO work reported in reference [5]. Two effective kinematic viscosities were retained, one for physical mixing and. the other for thermal mixing. The conservation laws were simplified for axisymmetric flow whereby all gradients in rotation about the centerline were not to zero, and all radial components of momentum were neglected as being small. Second axial derivatives and axial gradients divided by the radius were also neglected in momentum. Thermal radiative heat transfer and thermal generation were also neglected. In the derivation of the energy conservation and transfer functions the effects of the second viscosity, the square of axial gradients, dependence upon radial velocity, and gradients divided by the radius were minimized. The radial coordinate system was transformed into a stream function coordinate system after Von Mises. The transformation functions and the transformed conservation relations to which the finite difference procedure was applied are as follows:

Stream Function

$$\frac{\Psi \partial \Psi}{\partial \mathbf{r}} = \rho \mathbf{u} \mathbf{r} \quad ; \quad \frac{\Psi \partial \Psi}{\partial \mathbf{x}} = -\rho \mathbf{v} \mathbf{r}$$

wherein:
$$\frac{\partial \Psi}{\partial r} = -\frac{u}{r}$$
; $\frac{\partial \Psi}{\partial x} = \frac{v}{r}$

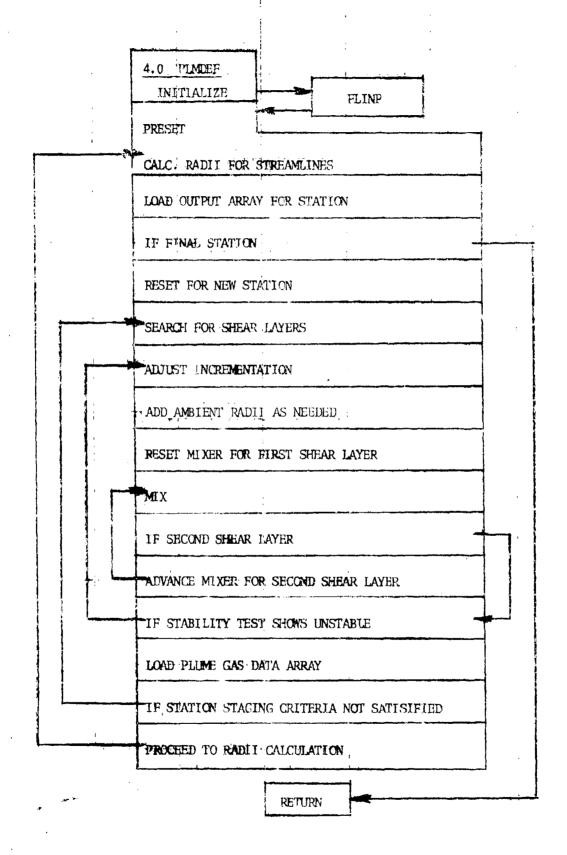


Figure 6. PLMDEF Flow Diagram

CONTINUITY:

for the stream function:

$$\frac{1}{2}\frac{3}{3}\frac{3}{3}\frac{4}{3} + \frac{3}{3}\frac{4}{3}\frac{3}{3}\frac{4}{3} - \frac{3}{3}\frac{4}{3}\frac{3}{3}\frac{4}{3} - 0$$

and after the transformation:

$$-\frac{\partial}{\partial x} \left(\frac{\Psi}{\rho u r} \right) + \frac{\partial}{\partial \Psi} \left(\frac{\rho v r}{\rho u r} \right) = 0$$

MOMENTUM:

$$\frac{\partial \mathbf{u}}{\partial \mathbf{x}} = \frac{1}{\Psi} \cdot \frac{\partial}{\partial \Psi} \left[\frac{\rho^2 \mathbf{u} \mathbf{r}^2}{\Psi} \varepsilon \cdot \frac{\partial \mathbf{u}}{\partial \Psi} \right] - \frac{\mathbf{d} \mathbf{p}}{\rho \mathbf{u} \mathbf{d} \mathbf{x}}$$

ENERGY:

$$\frac{\partial T}{\partial x} = \left[\frac{\rho^2 u r^2}{\Psi} \epsilon_T\right] \left(\frac{1}{c_p \Psi}\right) \left(\frac{\partial u}{\partial \Psi}\right)^2 + \frac{\partial}{\Psi \partial \Psi} \left(\frac{\rho^2 u r^2}{\Psi} \epsilon_T \cdot \frac{\partial T}{\partial \Psi}\right) + \frac{dp}{\rho c_p dx} + \left(\frac{\rho^2 u r^2}{\Psi} \epsilon_T\right) \left(\frac{1}{c_p \Psi}\right) \frac{\partial T}{\partial \Psi} \left(c_{p1} \frac{\partial \alpha_1}{\partial \Psi}\right)$$

DIFFUSION:

$$\frac{\partial \alpha_{\underline{1}}}{\partial \mathbf{x}} = \frac{\partial}{\Psi \partial \Psi} \left[\frac{\rho^2 \mathrm{ur}^2}{\Psi} \epsilon - \frac{\partial \alpha_{\underline{1}}}{\partial \Psi} \right]$$

where:

T = gas static temperature or thermal subscript

u = gas axial velocity

v = gas radial velocity

Prandt1 number = $c_p \rho \epsilon / k = 1$

Schmidt number = $\epsilon/D = 1$; in which D is the binary diffusion coefficient

E = effective kinematic viscosity

 α_{i} = species concentration

 $\frac{\partial c_p}{\partial r} \quad \text{is an assumed radial dependence for the gradient} \\ \quad \text{of the specific heat of a mixture of gases = } c_p/r. \\ \\ \text{Other quentities and terms are conventional.}$

The effective viscosity adopted for ASDIR includes the Donaldson and Gray treatment for the compressible free mixing of two dissimilar gases as reported in the ALAA journal, Vol 4, no. 11, in November 1966 and as applied by General Electric in SCORPIO-11 [5].

Overall, the plume model appears to reflect the assential features of measured plume data in those regions where the gas is hot. The regions where the model tends to lose accuracy are toward the outer edges where the gas shear appears to be too small thus allowing the plume to spread accessively. However, the plume temperatures and species concentration are essentially ambient in value so that the radiance and transmittance errors are small.

Viscosity Model

The choice of the turbulent viscosity, a, is critical to the plume flow field. Past models, available in the literature (Schlicting, Abramovich, Varren and Applied Science Laboratories), show the turbulent viscosity as a function of axial distance and velocities in the mixing region. Preliminary results for the flow field showed that the analysis using these models did not agree with jet engine test data. A modified temperature dependent model for viscosity was formulated to include the temperature effects of hot flow.

The model being used in this analysis is:

$$\varepsilon = x \cdot \{U_1 - U_2\}/[CNST \cdot (T_1/T_2)^2] \cdot \gamma$$

where

$$\gamma = 1$$
; $M_c < 0.6$
 $\gamma = 1.6/(1 + M_c)$; $M_c > 0.6$

The value for γ is from a paper by Donaldson and Gray. The value of CNST depends on the nozzle exit Mach number $(M_{\rm C})$ and the centerline temperature. (See Page A-85).

Stability and Print Criteria

In order to insure stability, the axial step size, Δx , must be kept small. It was found that the system became unstable when the influence parameter

$$cz = {\Delta x \cdot \varepsilon \cdot u \cdot r^2 \cdot p^2}/{2 \cdot \psi^2 \cdot T^2 \cdot r^2} \ge 0.20$$

To achieve a solution in a reasonable length of time, the step size Δx should be as large as possible. Accordingly, czm, the maximum value of cz for each axial location is examined. If czm is too great, Δx is reduced for ensuing calculations. If czm is too small, Δx is increased for ensuing calculations.

Flow parameters are not entered into the data array at every value of Δx calculated. The calculation step sizes are too small. Tabulation occurs whenever the number of Δx increments from the previous calculation exceeds NBS or the sum of the Δx 's exceed Δxp . Δxp is incremented after each tabulation such that

 $\Delta x p_N = x p_O (0.5N)$

Excessive static pressure at the nozzle exit plane is treated in a pseudo convergent-divergent nozzle manner which reduces the pressure to ambient value while accelerating and expanding the plume core to supersonic velocity over an axial distance of one nozzle diameter.

The program develops a plume gas data array containing the velocity, pressure, temperature, CO_2 concentration, H_2O concentration, and radius for up to 30 radii over an axial span of 49 stations. The velocity and pressure are multiplexed to share common registers. (See PLMDM for demultiplexing).

FLINP (Page A-87)

FUNCTION:

Provide a vehicle by which the required calculation parameters, nozzle exit flow quantities and data initialization can be read into ASDIR.

INPUT:

All control and computational variables in the PLUMIN namelist.

CUTPUT:

Computationally compatible engine nozzle exit plane flow quantity data. Nozzle and ambient data are printed.

SUBROUTINES:

CHEM, THRUST

DESCRIPTION:

The flow diagram of Program FLINP is shown in Figure 7. The function of this program is to initialize engine operational input and control quantities. The control function is derived from the input data. If the engine operational input quantities are not of the proper form, certain operations and additional reads are invoked to put the input into a form compatible with the PLMDEF program. Details can be found in ASDIR-II, Volume I, Users Manual.

In general, if PA is contained in namelist PLUMIN, the Mil Std 210 standard atmosphere in subroutine THRUST is bypassed. The inference is that an off-standard or test facility ambient condition is being simulated and the other ambient quantities are required in PLUMIN. Similarly, this program will bypass the default engine sample case if a nozzle exit velocity U8(1) is given greater than 1 ft per sec or if an ambient velocity UA is given greater than 11 ft per sec. Further, if U8(1) is greater than 1 the volume for T8T, P8, PQ, XCO2, XH20 will be accepted as given in PLUMIN. Again, if XCO2 has been omitted or given a small value (i.e. < XCO2A), the XCO2, XH2O, R, CAMMA, etc. will be calculated in CHEM for the fuel type (TANE) and stoichiometric equivalent ratio (EQR). Finally, if a gas quantity distribution for U8, T8T, XCO2, and XH2O is given in PLUMIN, they will be accepted unchanged since the load array key is taken from a test on U8(2).

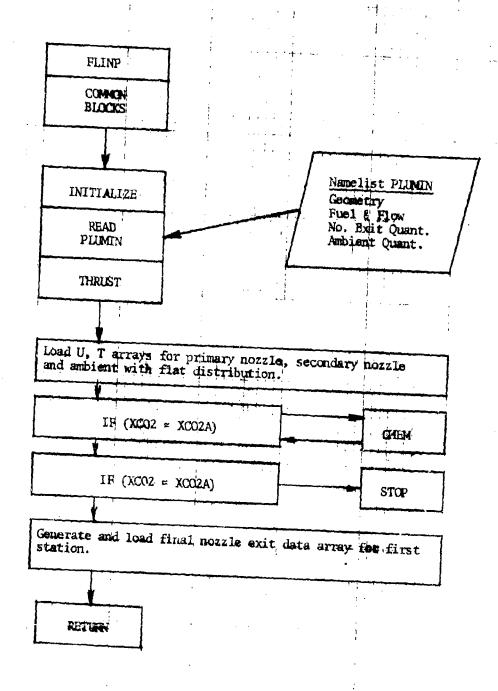


Figure 7. FLINP Flow Diagram

CHEM (Page A-91)

FUNCTION:

Generate the post combustion gas description (gas

constant, specific heat and specific heat ratio,

molecular weight, species conservation, etc) for a given fuel type, TANE, stoich ometric equivalence ratio, EQR.

INPUT:

TANE, FOR

OUTPUT:

Fuel to air ratio, gas constant, specific heat, gamma,

CO2 concentration, and H20 concentration.

SUBROUTINES:

None

DESCRIPTION:

This program calculates the gas properties for the products of combustion with ideal combustion. Since combustion products properties vary only slightly with pressure and tend to chemically freeze near 2500°F in typical nozzle flow, the precision of this simple college chemistry approach is adequate and the fast calculation

makes it a worthwhile approach.

THRUST (Page A-93)

FUNCTION:

Provide a vehicle by which the required calculation parameters and engine operating quantities can be read

into ASDIR.

INPUT:

All control and computational variables in the POWER

namelist.

OUTPUT:

Computationally compatible engine nozzle exit plane

flow quantities not provided in FLINP input namelist

PLIMIN. Flight condition data is printed.

SUBROUTINES:

M:HD

DESCRIPTION:

The flow diagram of subroutine THRUST is shown in figure

8. The function of this program is to read atmospheric,

engine type, data type code, and engine specific

performance data. The program control functions described for Program FLINP apply throughout this program.

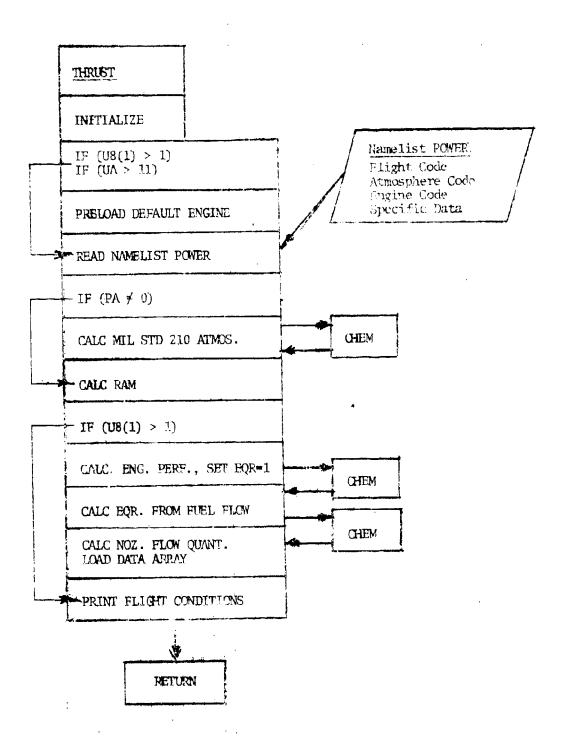


Figure 8. THRUST Flow Diagram

PLMOM (Page A-98)

FUNCTION:

To manage the plume gas data. An input control parameter, KDATA, controls the dispatch of the plume gas data.

INPUT:

Plume gas data and the control code KDATA.

OUTPUT:

Plume gas data output to cards, tape, printer, or printer plot in any combination.

SUBROUTINES:

PLMPLT, PLMPRNT

DESCRIPTION:

Each digit of the five digit integer KDATA controls an I/O function. KDATA is represented by the value:

KDATA = ABCDE

If A=0, a new plume will be calculated. If A=1*, an old plume previously filed will be brought in and plume calculations will be bypassed. If A>1, plume gas data will be read from input cards.

If B = 1*, the plume gas data array is filed in the computer for later or repeated analysis. If $B \neq 1$, the filing operation is bypassed.

If C = 1*, the plume gas data array is punched into output cards for later or repeated analysis. If $C \neq 1$, the punch operation is bypassed.

If D = 1, the plume gas data array is printed in output by subroutine PLMPRNT. If D \neq 1, the print operation is bypassed.

The subroutine PLMPLT is called with the final value E in the KDATA code. The E value has a different pattern than exhibited by A through D, see PLMPLT.

In addition to breaking down the code carried by KDATA, this program auds a layer of ambient data all around the plume data. The ambient data blanket is necessary to contain the interpolation exercises performed in the ray tracing calculations. The program then searches the plume gas data array to locate the tip of the core and

record such other parameters such as plume length AL, plume diameter REND, plume edge tangent TNB, etc. The minimum output RB, XC, REND, and AL is printed.

The PLMDM flow chart is shown in Figure 9.

^{*}This code represents a machine dependent operation and may require further development for satisfactory operation.

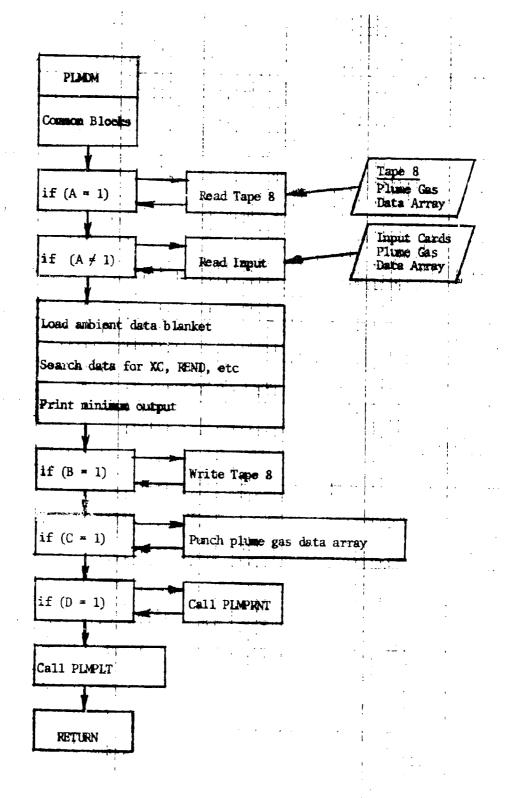


Figure 9. PLMOM Flow Diagram

PLMPRNT (See Page A-100)

FUNCTION:

Print the contents of the plume gas data array on

the output printer.

INPUT:

The plume gas data array

OUTPUT:

The plume gas data

SUBROUTINE:

None

DESCRIPTION:

Print the plume gas data array. The plume gas data is printed in the following sequence:

- general caption

- plume station number and location

centerline and edge Mach numbersradius, velocity, pressure, temperature, and CO2

and H20 concentration

Units are printed in the captions. The quantities printed are all self explanatory except the edge Mach number. The edge Mach number is that value at the outer edge of the innermost shear plane. This quantity will tend to oscilate due to the stepping and clipping process which occurs in the finite differencing procedure.

PLMPLT (See Page A-101)

FUNCTION:

Plot selected plume gas data quantities on the output printer.

INPUT:

The plume gas data array and the final digit of the KDATA control code.

OUTPUT:

Selected arguments of the plume gas data plotted on the output printer.

SUBROUTINES:

None

DESCRIPTION:

The output printer has 136 characters horizontally and more than 60 lines vertically. The grid represented by lines and characters are assigned one of the letters in the word plume (P L U M E) with the addition of # to permit the plotting of six ranges. A space is assigned the first character portion after a change in range. The nozzle exit plane value establishes the separator between # for values greater than the nozzle exit value and P for values between 80% and 100% of the nozzle exit value. Values in the 60% to 80% range are assigned L. Values between 40% and 60% are assigned U, etc. Each line represents a radius, and the interstation distance accumulates a number of characters. With two way interpolation a printer line is compiled and printed. After the data has been plotted, the axial scale is compiled and printed.

The code designating the plume gas data to be plotted is provided by the single digit value remaining in KDATA after having been stripped in program PIMDM. Up to four plots can be generated. In their plotting sequence, they are temperature, XO2 concentration, H20 concentration, and plume velocity. The appropriate captions are included. All four data arguments are plotted if the E value of KDATA is given as O. If E=1, no plots are printed. If E=2, the last three are printed, if E=3, the last two are printed, if E=4, the last plot only is printed, and if E is greater than 4, the program will fail. For the general breakdown of KDATA, see program PLMDM.

The final activity accomplished in program PLMPLT is the de-multiplexing of the velocity and pressure. The pressure is brought up from the back of each velocity-pressure register and replaces the velocity. The velocity is no longer retained in the plume gas data array.

ALPLUM (See Page A-47)

FUNCTION:

Calculate total apparent spectral radiance of exhaust plume and hot engine parts as received by an observer with plume and atmospheric emissions and attenuation, including background contributions

considered.

INPUT:

Flight conditions, engine hot parts and plume

parameters.

OUTPUT:

Engine/Plume IR Signature

SUBROUTINES:

RAYCAL, PLURAY, PLUSIG

DESCRIPTION:

The flow diagram of program ALPLUM is shown in figure 10. The prime function of this routine is to be the controlling program to allow calculation of aircraft infrared signatures. Description of the working programs controlled by ALPLUM are contained under the

individual program titles.

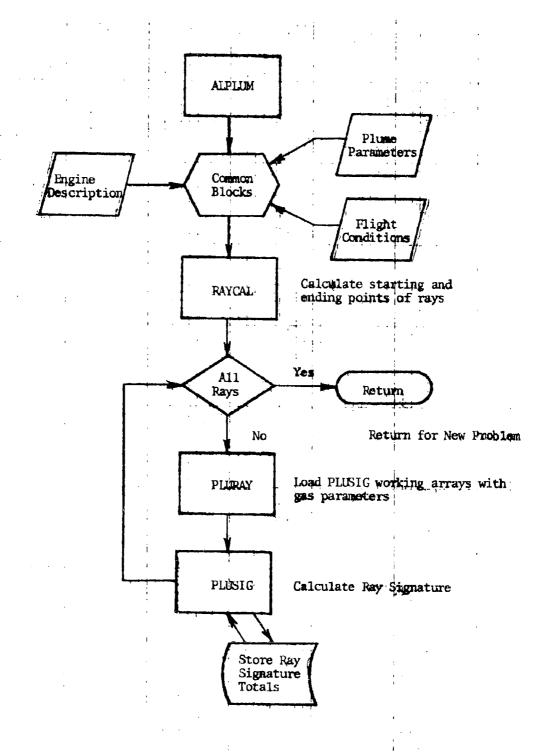


Figure 10. Total IR Signature Calculation.

RAYCAL (Page A-50)

FUNCTION:

To segment plume into discrete areas for signature analysis in other routines.

INPUT:

Plume and engine geometry.

OUTPUT:

Array of plume segment parameters.

SUBROUTINES:

None

DESCRIPTION:

RAYCAL, being basically geometric, is a relatively simple program. However, since signature prediction accuracy is dependent upon the geometric representation of the engine nozzle and the plume, graphical detail will be presented for this routine.

The engine nozzle and plume are characterized as a series of planar area segments each representing the termination of a ray or beam of radiated energy. Each segment represents either free (background) space or a black surface. The cross-sectional shape and area of each ray is defined by the segmentation scheme and is constant along the ray centerline. The segmentation scheme takes full advantage of symmetry as described with reference to the six accompanying figures.

Rectangular coordinates provide, figure 11, the framework for the ray definition. The X axis represents the engine nozzle centerline. The D axis, in the nozzle exit and perpendicular to the centerline X, is free to rotate about X so that the plane D-X will contain the centerline ray. The lateral axis, Z, is normal to both D and X. The angle of view, known as the aspect angle or ASPDEG, A8, is measured from the centerline X and lies in the D-X plane as shown in figure 12.

Dimensions on D represent the normal intercept of the ray centerline on plane D-Z from the Z axis.

Dimensions on Z represent the lateral intercept of the ray centerline on plane D-Z from the D axis.

Dimensions on X represent the axial intercept of the ray centerline on plane X-Z from the Z axis.

The simplified plume geometry adopted for use in this program is shown in figure 13. The plume is analyzed in three independent but adjoining regions:

- (1) The plume core is segmented in the X-Z plane into a number of radial and angular segments shown in figure 14. Segmentation begins at the extremity of the nozzle exit projection on the X-Z plane and continues to the extremity of the core.
- (2) The plume skirt is segmented in the X-Z plane into a number of radial and angular segments shown in figure 15. Segmentation begins at the extremity of the core and continues to the extremity of the plume.
- (3) The engine nozzle is segmented in the D-Z plane into a number of radial and angular segments shown in figure 16.

Each planar segment so generated also defines a ray segment. The D-Z or D-X ray segment centerline intercept is centrally located, see asterisks, in each planar segment. Further, each segment is considered to be the ray segments own projection on the D-Z and D-X planes from which the ray segment cross section area, RAR, is calculated.

The ray segments are generated on the first call to RAYCAL. Successive calls simply transfer previously generated data.

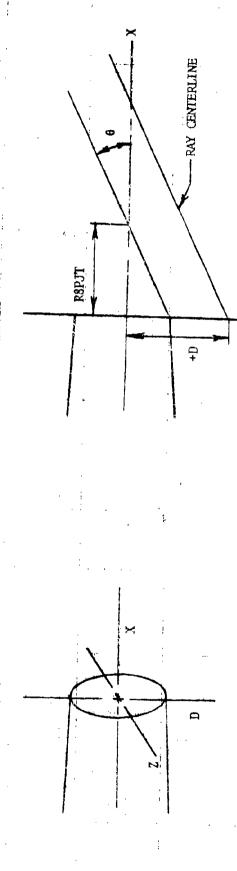


Figure 11. Ray Coordinate System

Figure 12. (D, X) Plane

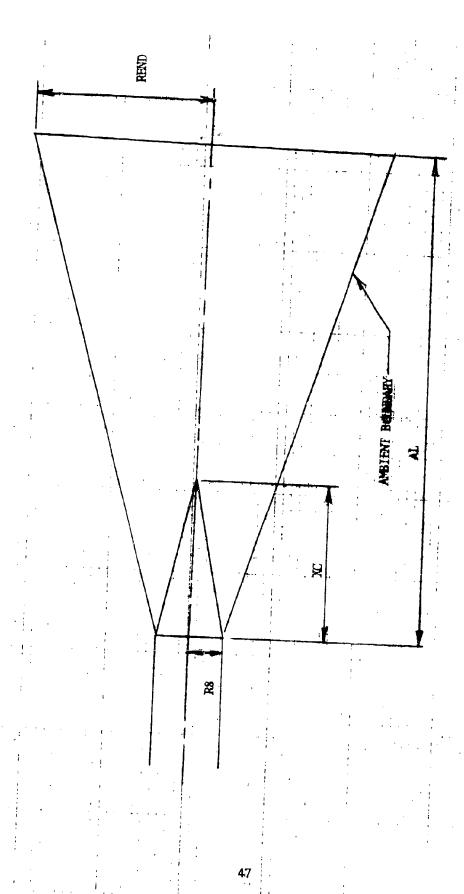


Figure 13. - Pitme Geometry

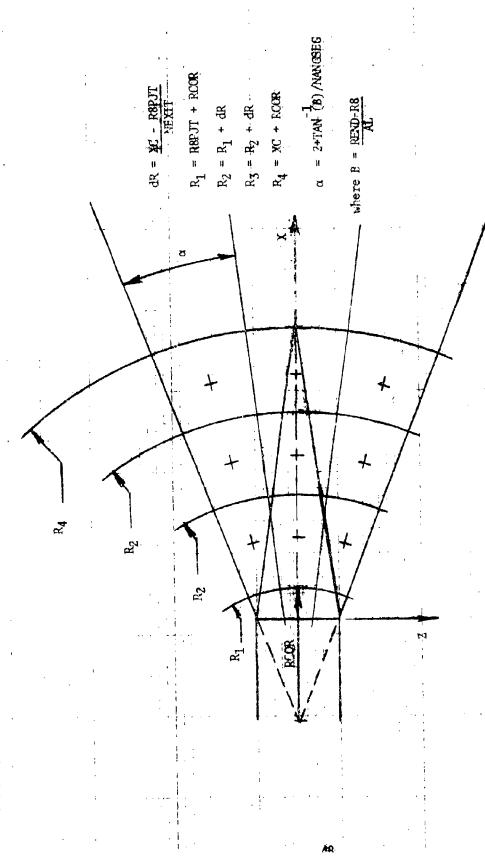
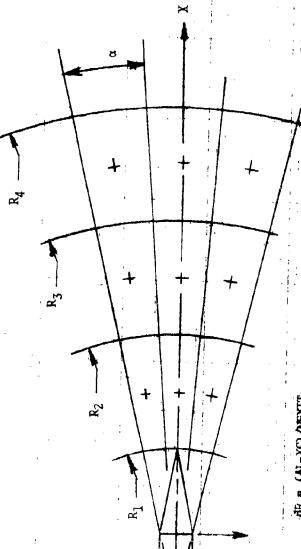


Figure 14. Segmentation of Core Region (X, Z) Pluma



dR = (AL-XC)/NEXET

$$R_1 = xc + kcoR$$

$$R_2 = R_1 + dR$$

$$R_3 = R_2 + dR$$

Segmentation of Non-Core Region (X, Z) Plume Figure 15.

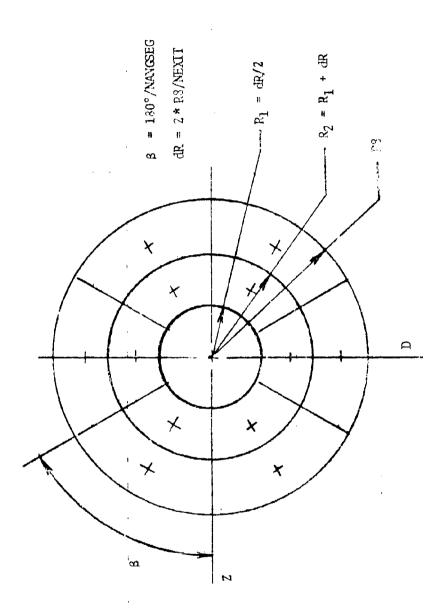


Figure 16. Nozzle Exit Segmentation

PLURAY (Page A-54)

FUNCTION:

Generate the parameters describing the ray which are required in the signature analysis routines.

INPUT:

Ray starting geometry, plume gas parameters.

OUTPUT:

Ray geometry, gas partial pressures and gas static

temperature arrays.

SUBROUTINES:

INTERP, START

DESCRIPTION:

By referencing the plume gas data and the ray definition parameters, this routine fills the pressure,

temperature, and length array with the proper data for ray signature analysis, see figure 17.

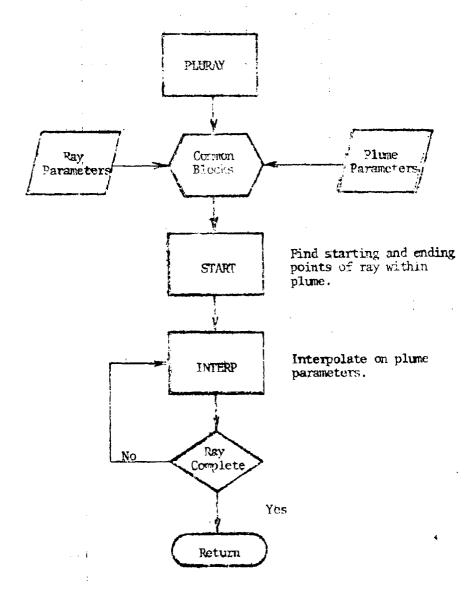


Figure 17. PLURAY Flow Diagram

START (Page A-57)

FUNCTION:

To calculate the starting and ending points of a ray through the plume.

INPUT:

Plume geometry, ray intersections

OUTPUT:

Starting and ending dimensions of ray

SUBROUTINES:

None

DESCRIPTION:

The starting and ending points are banded by engine geometry, engine plug geometry and flow field boundaries.

INTERP (Page A-61)

FUNCTION:

Interpolate on gas parameters to find values at a

given point whim the plume.

INPUT:

Gas parameters and coordinates of point in plume.

OUTPUT:

Concentration and temperature at input point.

SUBROUTINES:

Non

DESCRIPTION:

This program interpolates between stations along the centerline of the plume and along the radial dimension of the plume to find the static temperature, mole concentrations of water vapor and carbon dioxide and gas pressure at the input coordinate.

PLUSIG (Page A-63)

FUNCTION:

Evaluate the spectral signature and transmittance of a nonuniform gaseous path of NT segments.

INPUT:

Ray properties (temp, pressures and lengths), and frequency interval.

OUTPUT:

Spectral signature of gaseous ray and spectral transmittance of ray.

SUBROUTINES:

ATMOS, KBCAL, PLANCK, INTERPO, SETTAU, TAUCAL, TAUN20, ERF, DAH20, DCO2

DESCRIPTION:

The flow diagram for PLUSIG is shown in figure 18. The prime function of this program is to calculate the spectral signatures of nonuniform gas paths with various end conditions. The gaseous ray is divided into NT nonuniform segments with each segment defined by the partial pressures of $\rm H_{2}O$ ($\rm PH_{1}$), $\rm CO_{2}$ ($\rm PC_{1}$) and broadening gas ($\rm PN_{1}$), length ($\rm D_{1}$) and static temperature ($\rm T_{1}$). A brief discussion of spectral radiators follows:

Continuum radiation is obtained from liquids and solids because the molecules of the substance are in direct contact and exert forces on each other causing all levels of molecular emissions to occur. However, in the case of gas molecules, the intermolecular spacing is large and the free molecular vibrations and rotations result in discrete spectral emissions.

The intensity of these discrete gaseous emissions obeys the PLANCK radiation law as modified by Kirchoff's law. However, the emissivity of the gas is spectrally dependent and radiation is emitted or absorbed only in narrow wavelength intervals called spectral lines. These spectral lines are clustered into tightly spaced sets called bands. The emissivity of the gas is zero outside these bands and dependent upon line spacing and intensity within the bands.

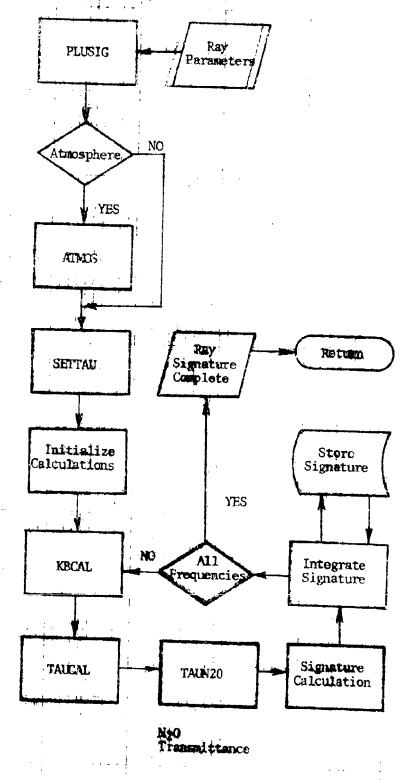
The inhomogeneous gas band-model expressions used in this program were developed at the University of Michigan and are based on the Curtis-Godson approximation. Rather than attempting to fully explain the excellent work performed at the University of Michigan Infrared and Optics Laboratory (now ERIM) in the areas of gaseous emission and absorptions, see references [5, 6, 7, 8, 9, 10].

Insert Atmosphere

Pressure Broadening Effects

Final : Gas Parameters

К & в



Figur : 18. PLUSIC Flow Diagram

A simplified discussion of the function of the program PLUSIG follows: A ray of NT segments, NA of which are atmospheric, of cross sectional area A is entered into the program, figure 19. Each of the NT-NA segments, representing the exhaust plume gas, are described by the gaseous partial pressures, temperatures, and lengths. The atmospheric segments are front loaded in the ray array through the use of the subroutine ATMOS and the range between, and altitudes of the observer and vehicle. The transmissions of each ray segment are calculated in PLUSIG using subroutines KBCAL, INTERPO, SETTAU, TAUCAL, TAUN2O, ERF and data contained in Data Blocks DAH2O and DCO2.

Figure 19 depicts the information available for a given frequency interval after the exercise of the transmittance calculation subroutines. The signature of the ray is found from

$$L_{ev_{ray}} = \sum_{i=1}^{NT} L_{ev}(T_i) \cdot \alpha_{vi}$$

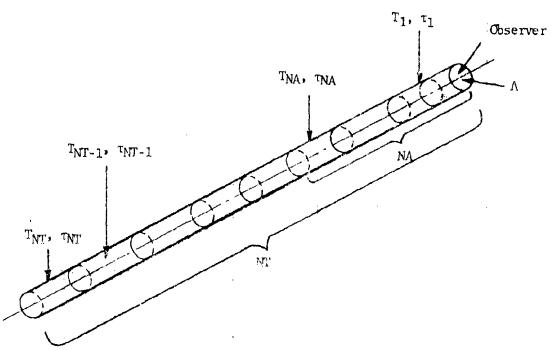
where $\boldsymbol{L}_{e_{\mathcal{V}}}$ is the PLANCK function and $\boldsymbol{\alpha}_{\mathcal{V}i}$ is the absorptance when

$$\alpha_{vi} = (1 - \tau_{vi}) \cdot \prod_{j=1}^{i} \tau_{vj}$$

The total transmittance, τ , for the gaseous path containing NT elements is extractable,

$$\tau_{\text{vay}} = \prod_{i=1}^{NT} \tau_{\text{vi}}$$

Thus the spectral transmittance can be used to determine the plume and atmospheric attenuation of a black surface at the terminus of the ray, figure 20.



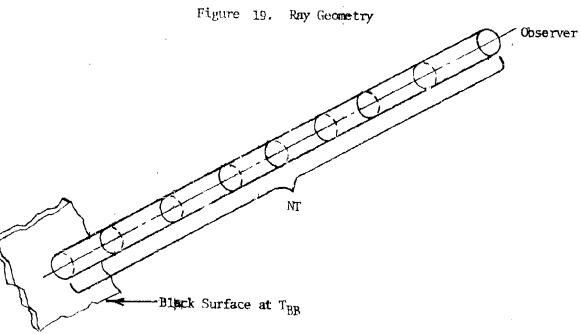


Figure 20. Ray Geometry with Terminating Black Surface

ATMOS (Page A-69)

FUNCTION:

Generate temperatures and partial pressures along an atmospheric path of NA elements and front load the ray pressure and temperature array with the

atmospheric data.

INPUT:

Range, altitude of plume and observer, number of

atmospheric segments

OUTPUT:

H₂O, CO₂ and N₂ Pressures

SUBROUTINES:

None

DESCRIPTION:

Given the geometry of figure 21, the altitude at point r is generated and the temperature and pressures are found from the empirical curve fits used in the computer code. References [8, 11, 12,

13].

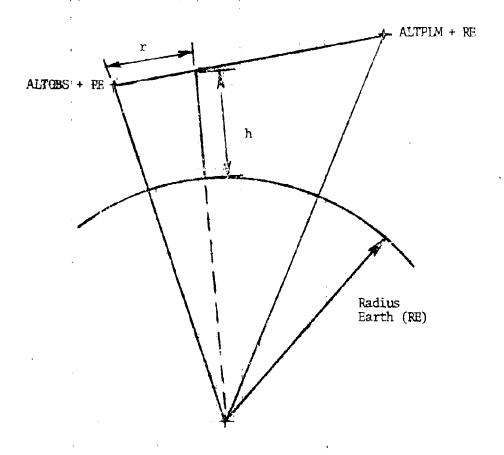


Figure 21. Geometry used in Atmospheric Model

SETTAU (Page A-70)

FUNCTION:

Insert pressure broadening effect of broadening gas by modifying effective gas pressures.

INPUT:

H₂O or CO₂ and N₂ gas pressures

OUTPUT:

Modified gas pressures (H_2O-N_2 and CO_2-N_2 mixtures)

SUBROUTINE:

None

DESCRIPTION:

References [6, 7, 8, 9, 10]

TAUCAL (Page A-71)

FUNCTION:

Calculate the gaseous transmittance in a given

frequency interval.

INPUT:

Pressures, temperatures, gos parameters (Υ & β)

CUTPUT:

Gaseous Transmittances, $\tau(v)$

SUBROUTINES:

INTERPO

DESCRIPTION:

Calculate transmittance of gaseous substance over a given frequency interval using overlapping line

approximations.

 $\tau(v) = \exp[-\beta_e \cdot f]$

Where β_{e} is the effective value of β and f is the Ladenburg-Reiche function. References [6, 10].

KBCAL (Page A-73)

FUNCTION:

To look-up and interpolate on the band model

parameters for a given frequency.

INPUT:

Frequency, gas type

OUTPUT:

K, β, temp and number of temperatures

SUBROUTINES:

None

DESCRIPTION:

The average line width to spacing ratio multiplied by $1/2~\pi$ called β , and the average line strength to spacing ratio called K are generated in this subroutine. These parameters are found at a given input frequency for H_2O and CO_2 over the temperate erange from $300~\rm K$ to $3000~\rm K$.

63

Function PLANCK (Page A-75)

FUNCTION:

Calculate PLANCK's blackbody function

INPUT:

Frequency (cm⁻¹), temperature (°K).

OUTPUT:

Watts/Steradian/cm³

SUBROUTINES:

None

DESCRIPTION:

From the classical expression for the radiant emittance into a Lambertian hemisphere

$$M_{ev} = \frac{2\pi (\kappa T)^{4}}{c^{2}h^{3}} \left(\frac{x^{9}}{\exp(x)-1} \right)$$

where

$$x = hv/kT$$

the spectral radiant sterance for a given frequency and temperature can be found $% \left(1\right) =\left(1\right) \left(1\right) +\left(1\right) \left(1\right) \left(1\right) +\left(1\right) \left(1\right$

$$L_{ev} = 2hc^2v^3/(\exp(x)-1)$$

See Reference [14].

Function INTERPO (Page A-76)

FUNCTION:

Interpolate over the N dimensional set (X, Y) to find the value at a given point (ARG).

INPUT:

X, Y, N, ARG

OUTPUT:

INTERPO

SUBROUTINES:

None

DESCRIPTION:

See Page A-76.

TAUN20 (Page Λ-77)

PUNCTION:

Calculate transmittance of $N_2\theta$ in the atmosphere over celected frequency intervals.

IMPUT

Frequency, atmospheric N20.

OUITUT:

Transmittance of $N_2{\rm O}$ in atmosphere.

SUBROUTINES:

ERF, INTERPO

DESCRIPTION:

See Page A-77.

Function ERF (Page A-78)

FUNCTION:

To evaluate the error function, erf(x).

INPUT:

X

OUTPUT:

ERF (x)

SUBROUTINES:

None

DESCRIPTION:

Evaluates the error function.

 $r \operatorname{erf}(x) \approx 2/\sqrt{\pi} \int_{0}^{x} \exp(-t^2) dt$

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APPENDIX

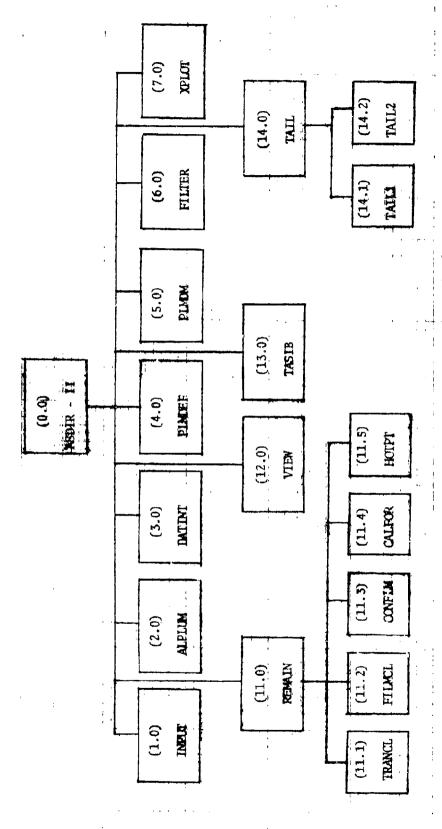
ASDIR-II

PROGRAM LISTING

APPENDIX

This appendix contains a listing of the Aeronautical Systems Division's Infrared Signature Predicting Model (ASDIR). The computer code was developed in the structural form shown in figure 22. The overlays are organized in three major functional groupings with a fourth overlay dedicated to data initialization. The prime functions of each overlay are:

Overlay	
(0,0)	Program Control
(1,0)	Data Input & Initialization
(2,0)	Plume Signature Calculations
(3,0)	Data Initialization
(4,0)	Plume Gas Dynamics
(5,0)	Plume Data I/O
(6,0)	Spectral Filter Data Initialization
(7,0)	Spatial Output
(11,0)	Hot Parts Temperature Calculation
(12,0)	View Factor Calculations
(13,0)	Hot Part Radiation Calculations
(14,0)	Not Part Data Control



Pigure 22. KSpik-II Overlay Structure

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130	160E-1, 156E-1, 1808E-1, 1808E-1, 1958E-1, 179E-1, 1068E-1, 1608E-1, 179E-1, 176E-1, 176BE-1, 179E-1, 176BE-1, 179E-1,		11.9
130 130	#\$0E-1## 976GE-1## 1418E-0;#166GE-0## 179GE-0## 145GE-0## 25GE-0##	:	119
13.6 13.6 13.6 13.6 13.6 13.6 13.6 13.6	E.1. 6850E-1. 3762E-11. 1.1.00 E-11. 1.1.00 E-11. 1.0.00		
123	E.1s.475GE_las 9634E-1s.1334GE-gas.299GE-gas.1250GE-gas.1250E-Gas.1250GE-gas.2591E-gas		
125 139 m	E-2,4000E-1,-7650E-1,-1120E-4,-1120E-8,-1130E-8,-1130E-1,-1120E-1,		121
139	E-2, 3520E-1, 6470E-1, 6470E-1, 1100E-1, 1100E-1, 1100E-1, 1200E-1, 1200E-1		221
13.5	E-2,,2520E-1,,5070E-1,,7650E-1,,8880E-1,,160E-1,,9500E-1,,9500E-1,,1600E-1,		7.51
13.5	E-2,4179DE-1,437DE-1,4546DE-1,4724JE-1,4828DE-1,49DUDE-1,4 E-2,4123DE-1,424GE-1,4443DE-1,468DDE-1,468DDE-1,468DDE-1,469DDE-1,464DDE-1,474DE-1,		
12.5 1.3.5 1.3.5 1.3.5 1.3.5	E-2.,1235E-1,.2946E-1,.4430E-1,.6080E-1,.6868E-1,.6860E-1,.4860E-1		
	E-21.0520E-21520E-1.12700E-15790E-15520E-15520E-1 E-31.6800E-21520E-12750E-14490E-15210E-1620E-1. E-31.4000E-21070E-12740E-13740E-1530E-1530E-1		921
	16-3,680GE-2,152CE-1,275GE-1,449GE-1,449GE-1,521GE-1,620GE-1, 16-3,400GE-2,1C7GE-1,214CE-1,4374GE-1,453GE-1,457GE-1,45		72
	31E-3r*4000E-2r,1070E-1r,2140E-1r,3740E-1r,4530E-1r,5330E-1r,		821
	THE PROPERTY OF THE PROPERTY O		129
E	こうじゅん こうじきこいき シストラフ・スタン・スタン・スタンド・スタン・スタン・スタン・スタン・スタン・スタン・スタン・スタン・スタン・スタン		021
£.	THE CONTRACT TO A CONTRACT TO		131
E	COMPANY OF THE STATE OF THE STA		(1)
	IOR-10-PRODESTANDARY SOURCESTANDARY CONTRACTOR CONTRACT		
	の日本では、人で行われている。他のものに「アメットの人の人に「アメント」というとしているのののでは、「アメントの人の人の人の人の人の人の人の人の人の人の人の人の人の人の人の人の人の人の人		25.
13.5	53E-3;.5446E-3;.3180E-2;.7560E-2;.2920E-1;.2040E-1;		**
	43E-3, 375GE-3, 185CE-2, 4633GE-2, 175GE-1, 4269GE-1, 4275GE-1,		135
	13E-3;•263CE-3;•119CE-2;•46GCE-2;•156UE-1;•253CE-1;•267OE-1;		135
कि व्यक्ति हो जिल्ला कि वर्ष को वर्ष का वर्ष को के वर्ष के वर के वर क क क क क क क क क क क क क क क क क क क	21F-4., 1850F-3., 9090E-3., 3600E-2., 1330E-1., 2410E-1., 2590E-1.		137
9 9 9 9 9 9 9 9 9 9 9 9 9 9 9 9 9 9 9	215 - L. 12705 - 3. 71125 - 3. 4165 - 2. 17705 - 1. 20505 - 1. 20506 - 1.		135
9 9 9			139
2 to 1	TOURING TO A CARREST TO STREET A CARREST CONTRACT OF THE CONTR		671
2 m de 1 m m m m m m m m m m m m m m m m m m	CTERTAR TO THE TOTAL	0440	
	NOTE THAT IN THE THAT IS NOT THE TOTAL TOTAL TOTAL TOTAL THE THE THAT IS NOT THE THE THAT IS NOT THE THAT IS NOT THE THE THAT IS NOT THE THE THAT IS NOT THE THE THE THAT IS NOT THE THE THE THE THE THE THE THE THE TH		6.7
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~	10E-4,,09900E-4,,2660E-3,,1670E-2,,6510E-2,,1560E-1,,2490E-1,	_	37
	30E-4,,1028E-3,,376GE-3,,167G-2,,644 <u>9E-2,,1528E-1,,2510E-1,</u>		145
4.42		_	9+1
	10 F 1 F 1 F 1 F 1 F 1 F 1 F 1 F 1 F 1 F	0	47
	AND THE STATE OF T		148
	・ 1 ・ 1 ・ 1 ・ 1 ・ 1 ・ 1 ・ 1 ・ 1 ・ 1 ・ 1		641
174.41	・ ・ ・ ・ ・ ・ ・ ・ ・ ・ ・ ・ ・ ・ ・ ・ ・ ・ ・		12.0
5. 7	TOTAL	, ,	
158	[<u>= </u>		
-	E-3, 2720E-3, 892CE-3, 4410E-2, 125CE-1, 2770E-1, 318UE-1,	_	26
]E-3.,326BE-3.,100BE-2.,49GE-2.,1470E-1.,2950E-1.,3433E-1.		153
. "	16-14. L2-10F-3. E450E-2. 5650E-2. 1610E-1. 3060E-1. 3750E-1.	_	• • • • • • • • • • • • • • • • • • • •
•	TO THE PROPERTY OF THE PROPERT		155
	Commonwealth Comm		156
155 1.3620	**************************************		
1.50	1E-31_65900E-31.27.60E-21.68880E-22.23.40E-1.68.310E-7.00JE-4.1		
	E-3. 1180E-2. 3220E.2. 1180E-10.2620E-1. 4510E-1. 5500E-1/	_	261
2	COFFE		159
,520	15-41. 16605-2. 16606-2. 12605-1. 12605-1. 29205-1. 29205-1.		291
	15.25 23475 22 34476 22 4446 6 24 4400 6 5 1 5 1 5 1 5 1 5 1 5 1 5 1 5 1 5 1 5	_	161
: '	THE CONTRACTOR IN THE PROPERTY OF STREET STREET STREET STREET		791
,	THE TRANSPORT OF THE PROPERTY		
1.26	12 . Z.P. S.U.U.E Z.E. D.C. D.C Z.E. Z. S.U.E Z.E. Z.E Z.E. D.C. G.E. C.E. C.E. S.E. Z.E Z.E. Z.E. Z.E. Z.E. Z.E. Z.E.		
1.29	1E-2, 3300E-2, 7010E-2, 2030E-1, 4600E-1, 7820E-1, 8650E-1, 9	٠,	•
1.310	1E-2, 3700E-2, 9460E-2, 2200E-1, 5190E-1, 8090E-1, 9700E-1,		601
•	F=2,_6000F=2,_9693E=2,_2390E=1+,6620E=1+,1390E=0+,1060E=0+	_	991
	F=2. 45007=2.1410F=12720E=12720E=04090E=04170E=0	0	791
- 0	11 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1	0	168
	THE CONTRACT TO A CANDELL ASSETTANT TO SECURE	_	164
	COLUMN TO THE CO		170
**************************************	TOTAL COMMENT OF THE PROPERTY	•	121
178	1011-12	¦	

14	5351E-10.5000E-10.3790E-0.2760E-0.2700	216 216 225
1306-1 0AH20 216 11906-1 0AH20 200 11756-0 0AH20 201 12766-0 0AH20 201 12766-0 0AH20 202 12030-1 0AH20 203 13006-1 0AH20 206 13006-1 0AH20 207 15006-1 0AH20 207 15006-1 0AH20 211 14906-1 0AH20 211 14906-1 0AH20 215 12906-1 0AH20 216 13906-1 0AH20 226	18-04.5000E-18.3790E-04.2766E-04.270E-8.1920E-05.2000E-04.200E-04.200E-04.200E-04.200E-04.200E-04.200E-04.200E-04.200E-04.200E-04.200E-04.200E-04.200E-04.200E-04.200E-04.200E-04.200E-04.200E-04.200E-04.1950E-04	
1350E-0 0AH20 197 1540E-0 0AH20 193 1540E-0 0AH20 201 1775E-0 0AH20 201 1276E-0 0AH20 201 1276E-0 0AH20 205 1500E-1 0AH20 205 1500E-1 0AH20 207 1500E-1 0AH20 207 1500E-1 0AH20 211 1490E-1 0AH20 211 1490E-1 0AH20 211 1490E-1 0AH20 211 1490E-1 0AH20 215 2350E-1 0AH20 215 2350E-1 0AH20 215 2450E-1 0AH20 215 250E-1 0AH20 225 1190E-1 0AH20 223	10-0. 5003E-0. 3790E-0. 3150E-0. 2270E-3. 1920E-0. 2050E-0. 2050E-	3 5
.1350E-0	18-94,5003E-18,3790E-0,2766E-18,270E-18,21920E-18,200E-18,21920E-18,200E-18,21920E-18,200E-18,	216
1350E-0 0AH20 295 1550E-0 0AH20 291 1750E-0 0AH20 201 1750E-0 0AH20 201 1270E-0 0AH20 205 1607E-0 0AH20 205 1607E-1 0AH20 205 1507E-1 0AH20 206 7107E-1 0AH20 210 1507E-1 0AH20 211 1407E-1 0AH20 215 1207E-1 0AH20 225	10-0.50000000000000000000000000000000000	28 5 16
1350E-0 0AH20 201 1770E-0 0AH20 201 1770E-0 0AH20 201 1770E-0 0AH20 201 1890E-0 0AH20 205 1610E-0 0AH20 205 1610E-0 0AH20 205 1610E-1 0AH20 206 7100E-1 0AH20 206 7100E-1 0AH20 210 8500E-1 0AH20 211 8500E-1 0AH20 215	10-0. 500 mm - 1. 400 0E - 1. 400 0E - 0. 20 20 E - 0. 19 20 E - 10. 19 20 E	5 5 6
.1350E-0	10-4; \$603E-E; 3799E-Q; 2766E-B; 2309E-B; 21920E-I; 3000E-P; 3790E-Q; 2905CE-B; 2306E-D; 2006E-P; 2006	15
1350E-0 DAH20 195 1750E-0 DAH20 201 1750E-0 DAH20 201 1750E-0 DAH20 201 1270E-0 DAH20 202 1203E-0 DAH20 205 1610E-0 DAH20 205 1610E-1 DAH20 205 1750E-1 DAH20 206 1710E-1 DAH20 210 1750E-1 DAH20 211	59516-0. 50006-1. 37906-0. 27606-0. 22306-0. 20206-0. 20206-0. 31506-0. 23306-0. 20206-0. 20206-0. 31506-0. 23306-0. 202	[5 16 1 1 1 1 1 1 1 1
1350E-0	59:16-4, 50006-1, 37906-0, 27606-0, 23706-0, 2306-0, 2	216
1360E-0	\$\$\\\ \text{\$\frac{1}{2} \$	16
1350E-0	5951E-0: 5000E-0: 3790E-0: 2760E-0: 2760E-0: 2300E-0: 200E-0:	216
1350E-0	\$\$\\\ \text{\$\frac{1}{12\triangle}} \ \text{\$\frac{1}{12\trian	16
.1350E-0	5951E-0, 5000E-0, 3790E-0, 2760E-0, 230E-0, 230E-0, 1920E-1, 1920E	6
.1350E-0	59:18-4, 50008-10, 37908-0, 27608-N, 22708-3, 19208-16:16:16:1, 30008-10, 37908-0, 23008-0, 23008-0, 2008-16:16:16:16:16:16:16:16:16:16:16:16:16:1	6
1350E-0	59:JE-0, 5000E-0, 3790E-0, 2760E-0, 230E-0, 2420E-1, 1920E-1, 6613E-1, 3600E-0, 3590E-0, 23150E-0, 2430E-0, 2200E-0, 2300E-0, 230	
1350E-0 DAH20 1540E-0 DAH20 1540E-0 DAH20 1770E-0 DAH20 1770E-0 DAH20 1770E-0 DAH20 1700E-0 DAH20 1700E-0 DAH20 1700E-1 DAH20	\$\$\\\ 1706-6.4000E-0.3799E-0.3150E-0.2394E-0.2394E-0.2020E-1.202	
.1360E-0	5951E-0: 5000E-0: 3790E-0: 2760E-0: 2270E-0: 21920E-1: 1920E-1: 19	
.1360E-0	59; JE-0, 5000E-0, 3790E-0, 2760E-0, 2270E-0, 1920E-0, 4000E-0, 4000E-0, 3150E-0, 2270E-0, 2000E-0, 20	
20 - 0	5951E-9, 5000E-0, 3790E-0, 2760E-0, 2270E-3, 1205E-1, 200E-1,	
20 - 0 AH20 20 - 0	59516-4:50016-0:37996-0:37606-0:2706-0:32706-0:2308-3:3298- 11736-0:40006-0:4506-0:31506-0:24306-0:2206- 46136-1:3006-0:3586-0:23506-0:2306-0:2006- 18306-1:2566-0:3566-0:23506-0:3506-0:2006- 73606-2:3566-0:3666-0:3756-0:3506-0:3506-0:3506-	
20E-0 0AH20 00H20	5953E-4:5009E-E:3399E-E:350E-0:3150E-0:228E-8:328E-8:328 1173E-0:4000E-E:339E-0:3150E-0:2430E-0:31 4613E-1:3000E-0:388E-0:239E-0:3190E-0:30 1833E-1:3000E-0:385E-0:2359E-0:1950E-0:31 1833E-1:3000E-0:385E-0:2359E-0:1950E-0:31	7
20E-0 0AH20 60E-0 0AH20 60E-0 0AH20 70E-0 0AH20 70E-0 0AH20 70E-0 0AH20 70E-0 0AH20 70E-0 0AH20	595,1E-4,,5009E-E-,3799E-E-,760E-E-,2780E-H-,1920 1170E-6,,4000E-E-,4230E-0,3150E-4,2740E-4,2740E-4,2720 1667,74,300E-1,4350E-E-,2750E-E-,2750E-4,2740E-1,2720E-4,2740	•
20E-0 DAH20 60E-0 DAH20 00430 00430 70E-0 DAH20 70E-0 DAH20 70E-0 DAH20 70E-0 DAH20 70E-0 DAH20 70E-0 DAH20	52515-2:50035-5:37905-0:37605-3:2706-3:21206-3:21006-3:21006-3:21006-3:21006-3:21006-3	
20E-0	100261.41.100306.42.100375.40.40.40.40.40.40.40.40.40.40.40.40.40.	
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	E-4. 1920E-3. 4700E-3. 4670E-2. 1660E-1. 3130E-1. 1700E-4.	DAHZO	237
	E-4 2600E-3 1658E-2 5668E-1 1858E-1 3418E-1 3998E-1/	2	238
	1 COEF11/	ç	523
	2]E-6,,2]3QE-3,,129QE-2,,736QE-2,,229QE-1,,378GE-1,.422GE-1,	~	N
	10E-43, 3060E-34, 1830E-21, 9320E-22, 18399E-11, 1460E-11, 1460E-11, 1	Υ.	1+2
	5.7E-43995E-32465E-21286E-13026E-14366E-1456CE-1.	0 AH20	242
	•1131E-13.e193E-3. 3462E-2.•1610E-1. 3560E-13. 45392-13. 4653E-14.	77	243
	7636-3. 82536-3. 46406-7. 20006-1. 464766-1.	AH2	542
	CONTRACTOR OF THE STANDARD OF	AH2	245
-	105 mm - 20 mm - 2 mm -	AHZ	246
	17.47 F 17.47	DAH23	247
	74.415 - 12.415 - 14.445 - 14.	DAM20	248
	ODDITION OF THE CONTRACT OF TH	DAHZO	672
	DEFINE AND CONTROL OF A CONTROL	DAHZO	250
	12. E44. 7300	DAH20	251
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	2月~17~17~17~12日のは、17~17~17~17~17~17~17~17~17~17~17~17~17~1	DAHZO	255
	TO SELECT THE SELECT T	04420	256
	・ 19 ののでは、 19 のののでは、 19 でものできないできない。 19 ののできないできない 19 のののできない 19 ののできない 1	DAKZO	257
	17日本のフラインの「大きのフラフト」というできます。 ちゅうこうじょ マイ・ドライン トー・・・・・・・・・・・・・・・・・・・・・・・・・・・・・・・・・・・・	DAHZO	259
	N. TOTAL CONTROL OF THE CONTROL OF T	DAMED	259
	2850F-1. 3650E-1. 4230E-1. 04610E-10-4820E-10-4500E-10	DAHZO	260
	11.11.11.11.11.11.11.11.11.11.11.11.11.	DAM20	261
	-8-4220F-14-5890E-14-5980E-1-5720E-1-540E-1-4700E-1-	DAHZO	262
	JE-8.1850E-3.8440E-1.66870E-1+.5930E-1,.5600E-1.8600E-1	DAMEG	263
	E-1,.7180E-1,.6630E-1,.6180E-1,.5560E-1,.534CE-1,.4780E-1,	DAMED	. 192
	E-1.0.44332E-1.0.5793E-11.0547GE-14.0563GE-1.0.4950E-1.0.4950E-1.0.4603E-1.	DAHED	265
	98578-157538-158908-158108-145108-145108-144908-142508-1-	DAH20	592
	F-1,. E320E-1,. 5390E-1,. 6400E-1,. 4540E-1. 4466E-1, 4000E-1,	DAHZO	267
	6887E-1,.6830E-1,.5483E-1495CE-1,4463CE-1,,458UE-1,,4035E-1,	DYHSO	263
	5200E-19.5151E-10.4830E-14490E-14540E-104180E-1.	DAHZO	269
	1588E-1 3508E-1 4518E-1 4510E-1 4648E-1 4528E-1 4450E-1 4490E-1	DAMED	270
	6203E-2, 2330E-1, 3590E-1, 3690E-1, 3600E-1, 96160E-1, 6170E-1, 4620E-1,	04420	27.1
	2701E-2, 1580E-1, 2220E-1, 339CE-1, 3660E-1, 3840E-1, 4060E-1,	CAHEO	212
	E-2, . 1810E-1, . 2030E-1, . 2630E-1, . 3839E-1, . 3338E-1, . 3608E-1,	DAHZO	273
	91E-35930E-21480E-12360E-12470E-12950E-13200E-1.	CANCO	*12
	553E-3, 3160E-2, 3E90F-2, 1540E-1, 2030E-1, 2560E-1, 3660E-1, 2560E-1, 2560	04450	27.5
i	481E=31381E+25890E-2120E-1		974
	543 E-32 - 600 E-32 - 627 GE-24 - 850 GE-28 - 1360 E-12 - 130 GE-28 - 130 GE-13 - 130 GE-1	0400	27.0
	3]E-3,.2625E-3,.2685E-7.595E-7.5950E-7.5950E-7.50E-7.50E-7.50E-7.70E-7.7	0 2 7 7 0	270
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	CORPARATOR AND	2440	
	THE FORM MANUAL STREET	0.2470	200
	#FOXTONET#************************************	DAH20) m
	0E-43	DAHZO	284
		•	
		744	285

340			335	•			300				325	} !			60.00					07		J		Cal 100 100 100 100 100 100 100 100 100 10			************************************	40,1			(c)				622	305	The state of the s			162				
0E-1,.8930E-1,.435CE-1,.255CE-1,.1990E-1,.19 ² 0E-1 3E-1,,6830E-1,.3780E-1,.239CE-1,.195CE-1,.1920E-1 0E-1,.473JE-1,.3643E-1,.238DE-1,.1970E-1,.1926C-1	3233E-1;47400E-1;44490E-1;4261fE-1;4214fE-1;2221fE-1;422060E-	1,.270CE-1,.229CE-1,.243CE-1,.	1 U U	1320E-1,.2710E-1,.4020E-1,.2940E-1,.2740E-1,.2820E-1,.2830E	20E+1, 2980E+1, 3180E+1, 29\$0E+1, 2950E+1, 2950E+1, 2990	7:016:2,,48266:1;,48466:1;,29766:1;,29766:1;,30666:1;,48766:1;	SOOM-No ANDRONES AND SOMETHIS ADDRONES, SOUTHIS AND SOMETHIS AND SOMETHIS AND SOUTHIS AND	\$501E-2, 1060E-1, 0272CE-1, 0279CE-1, 0204CE-1, 0317CE-1, 0317CE-1	DE-2, 9200E-2, 2380E-1, 273CE-1, 250DE-1, 304CE-)E-2, 8800E-2, 220FE-1, 253FE-1, 272FE-1, 30FFE-1, 30FFFE-1, 30FFE-1, 30FFFE-1, 30FFFE-1, 30FFFE-1, 30FFFE-1, 30FFFE-1,	29.36.4257366654.43666537665376666574534664534566666666666666666666666	50E+2; 505E+2; 114EE+1; 145E+1; 155E+1; 25EE+1; 27EE+1; 27EE+1	03E-2, 43CDE+2, 929CE-2, 83DE+1, 233CE-1, 234	1850E-2, 3700E-2, 9C70E-2, 168CE-1, 220E-1, 292VE-1, 3290E	21.3310E-21.813EE-21.161EE-11.2120E-11.288CE-12.3200E	\$_\\\\\\\\\\\\\\\\\\\\\\\\\\\\\\\\\\\	THE COCTACT OF THE PROPERTY AND THE AUTHORISM STRONG	0.00	810E-3, 1900E-2, 5370E-2, 1160E-1, 1790E-1, 2540E-1, 2870F	CONTRACTOR AND	7.435413	1.256)27.3, 41002422	1650E-3, 6723E-3, 2653E-2, 6410E-2, 1310E-1, 2110E-1, 2330	3,,2170E-2,,5490E-2,,1240E-1,,2050E-1,,222	4)415-3. 59155-3. 17455-7. 1855-7. 1857-7. 1857-7. 1857-7. 187	31.1225E-2, 3590E	3, 28805-2, 92205-2, 17365-1, 1850	::::::::::::::::::::::::::::::::::::::	1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1	0.00 t ** 1.00 0.00 0.00 0.00 0.00 0.00 0.00 0	10E-3, 165CE-2, 602CE-2, 131CE-2, 1413CE	THE MANAGER OF THE STATE OF THE	ARE CONTRACT HOUSE IN A CONTRACT OF THE PROPERTY OF THE PROPER	1,13116-4,,61206-4,,2776E-3,,11306-2,,4650E-2,,1100E-1,,1259E-1/	E-3; 1110E-2; 4370E-2; 1040E-1; 1193		111	3, 83205+3, 31705+2; 84205+2, 190	.2850E-27760E	E-3. 8050E-3. 2600E-2. 7320E-2. 97	3,.8610E-3,.2470E-2,.6910E-2,.9590	47-47-51-52-18-0-18-0-18-18-18-0-18-0-18-0-18-0-18	4.300777, FB30747, 3266743. 3266763. 3120763. 32200742. 194777777	TINGES OF
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177E-1,. 3690E-1,. 3650E-1,.2490E-1,.2120E-1,.2040E-1,.2000E-1,	AH20 344
100E-1, 1700E-1, 4190E-1, 2720E-1, 2280E-1, 2130E-1, 2240E-1,	***
1988 - March Control of the Control	- ·
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955E-11.5655E-12.575E-1.585E-1.685340E-14.6833E-14.68545E-14.6845E-14.685E-14.	
701E-1, 0100E-2, 1910E-1, 1750E-1, 1810E-1, 1940E-1, 2100E-1,	_
013E-11- 3700E-24-1050E-11-1276E-1-15520E-11-1710E-1-1400E-12-	
4516-1, 17006-2, 5406-2, 65506-2, 65506-2, 14306-1, 13166-1, 15006-1, 15006-1,	
24.17F-3. 60306-3. 30106-2. 65306-2. 33306-2. 93406-2.	
75JE-3.4100E-3.1930E-2.3660E-2.1190E-2.3320E-2.500EE-2.	_
45)E-31,2890E-31,1310E-2,,2320E-2,,2470E-2,,2560E-2,,4200E-2,	
)10E-3,,1600F-3,,9150E-3,,1500E-2,,1860E-2,,1970E-2,,3700E-2/	_
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943E-4.0100E-8.05650E-7.05650E-7.04840E-7.05050E-7.0508E-7.050	
7. JE - 4. p. 7. 7. 7. 7. 7. 7. 7. 1. 4. 7. 7. 4. 4. 7. 7. 6. 4. 7. 7. 7. 7. 7. 7. 7. 7. 7. 7. 7. 7. 7.	
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135-4. 3610E-4. (1220E-3. 443FCE-3. (124CE-2. 1620E-2. (290E-2.	
201E-4., 2450E-4., 1210E-3, 4350E-3, 1180E-2, 1670E-2, 2913E-2,	
101E-5.1700E-4.1030E-31.4390E-3.1260E-2.1920E-2.2550E-2.	
531E-5, 1200E-41, 1660E-41, 267CE-31, 1190E-21, 195E-21, 190E-21, 190E-21, 190E-21, 190E-21, 190E-21, 190E-21,	
40)E-5, 4000E-5, 7160E-4, 751CE-3, 7160E-3, 1260E-2, 1260	
アナンチャロンロスドーンチャンシンドーチェンスンチンド・シャットレインド・アチャンコンド・アナインド・アード・アード・ア・ア・ア・ア・ア・ア・ア・ア・ア・ア・ア・ア・ア・ア・ア・	
2251E-5, 9240E-5, 3240E-5, 3250E-4, 2520E-3, 1160E-2, 2150E-2, 3301E-2	
233E-5,.6430E-5,.3710E-4,.2680E-3,.127CE-2,.2110E-2,.3400E-2,	_
2316-51,9210E-51,3960E-41,2730E-31,1280E-21,260E-21,3550E-21	
2350f-50.1830f-40.4830f-50.2830f-50.2830f-30.1820f-20.2820f-30.	
70JE-5. 1800E-4. 1200E-3. C40CE-3. 1670E-2. 2510E-2. 4100E-2.	
835-5,,2080E-4,,9870E-4,,5890E-3,,1718E-2,,2570E-2,,420DE-2/	AH2) 376
[4 COEF18]	
*****CACCCC***************************	. ~
015-5, 3000E-4, 2040E-3, 6840E-3, 1840E-2, 2850E-2, 4580E-2,	
280E-5. 3300E-4. 2760E-3. 8190E-J. 1990E-Z. 2970E-Z. 4790E-Z.	
1635-6, 38306-6, 31706-3, 85906-40, 23666-2, 00806-2, 4506-2,	ATIVO MOS
5JE-4-4-4-700 -4-4-6-400E-34-4 0400E-3-4-240E-2-2930E-2-4990E-2-	
-4. 5600E-4. 3010E-3. 9410E-3. 2430E-2. 3420E-2. 5000E-2.	
383E-4.,5330E-4.,2800E-3,.1678E-2,.2646E-2,.3530E-2,.5516E-2,	•
30/16-4, 71006-4, 27656-4, 27656-17, 10936-2, 27686-7, 39786-7, 30786-7, 50486-2,	
#844_F***********************************	
015-4, 95005-4, 1710E-3, 1350E-2, 3060E-2, 3040E-2, 6990E-2,	_
930E-4, 9800E-4, 4340E-3, 1470E-2, 3150E-2, 4050E-2, 4950E-2	7
101E-6. 9996E-6. 3970E-3.1430E-2. 11880E-2. 3840E-2. 4990E-2.	
JE-4, 9800E-4, 7640E-3, 1410E-2, 3170E-2, 3870E-2, 3810E-2, 460JE-2	
DOMEN, OFFICE AND MODEL BOOK AND CONTROL OF THE CON	
E-31,1434E-21,1310E-21,2372EE-21,455E-21	DAH20 398
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[-2, 14611-2, 1500E-2, 1500E-2, 1601E-2, 1600E-2, 1600E-2, 1601E-2, 1601E-2	-2.130E-2.150E-2.150E-	100 mm m m m m m m m m m m m m m m m m m	4444	-3, 9500:-3, 1270E- -3, 1900E-7, 1260E- -3, 1820E-2, 1260E-				# # # # # # # # # # # # # # # # # # #		-2, 2976 -2, 2956 - -2, 2846 -2, 3236 - -2, 2866 -2, 3236 - -2, 2576 -2, 29536 - -2, 2466 -2, 25536 -	-2.3530 -2.528 -2.3590 -2.590 -2.3510 -2.3950 -2.3510 -2.3950 -2.3520 -2.3950
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CCECUL 55 (1.,25907 +1, 9830E+D, 431EE+1, 6 76 (1.,303TE+1, 4113F+1, 4801E+1, 6 46 (1, 4357w2 +6, 4271F+1, 5337E+1, 4	[+1; -2]]][[+1; -7]] [+1; -2]][[+1; -4]]	[11] - 2333[+ 1	7	E+1,.29665+5,.3950E+0,.20475+1+. 5+1,.19845+5,.4537E+0,.227*E+1,.	E31; 23616+L9+3C256+C9+18396+19+	. T + 1 / 1 / 1 / 1 / 1 / 1 / 1 / 1 / 1 / 1	E+G, 2787E+u, 5776E+D, 1526E+1++	E+0,,1289F (4,, 3812F+0,,1273E+1,,		**************************************	E + 22 + 5556 E + 22 + 2145 E + 24 + 5	E-20, 4311E+6, 46984E+C+41265E+1++	C+C, - C5035+Q, 28245+Qz - 125555+Lz.	- +Ook のの対象的すの。そのCOEの中の。それは自身の中性のでのCOEのCOE			E41, 1176E 1, 1332E	E+1+: 2027F+1: 12427c	56 + 0 + 123 EE + 1 + +		5030++++65538+0: 1273E+1++	5 - 41975 + 5 - 45092 + 5 + 5 + 4075 + 407	てい、それのないのであり、そのないのできるような事情の見りをはまる。 こういき かんかん かいしょう かんかん はいしょう はんのいけんきゅう	E+2++51505+2++1079E+1++			*\$ 9E + Q _ \$ 888 E + C + 1	454054110654144	いっぱん 一番自己を行って、「カリカロのなっ」とも、日本の自己を行った。 じょうじゅうしゅう かいかいかい かいかい かいかい かいかいかい かいかいかい かいかい か		17、 On a Man Man Man Man Man Man Man Man Man	7721	7+2++ 39325+6: - 6217E-0++ 1504E	THE PARK SCHOOL OF STREET OF STREET		STEEL STREET OF STREET OF STREET	10. 44256510. 7009610. 150005414.	Er., . 65575+0 7225E+0 17795+1	70 - Co. 460 5 6 6 4 0 2 4 7 4 5 6 6 7 4 2 4 4 4 6 6 6 6 4 4 4 4	FeG. 4970F+0 - 7729E+0 - 1965E+1-
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C NEN=2	NEM=2		TNPUTS	135
New New	### ### ### ### ### ### ### ### ### ##		TNPUTS	136
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DO 210 [13.4] Ja (1 + 1) / 2	280 KEN1J) = KENREN(1) Z10 CONTINUE Z10 CONTINUE Z10 CONTINUE Z10 CONTINUE Z10 CONTINUE Z10 CONTINUE RF=RP		TUPUTS	85.4
1	220 REN1) = XENREN(1) 220 CONTINUE 220 CONTINUE 220 CONTINUE 220 RENT (1) = XENREN(1) 220 CONTINUE 230 CONTINUE 230 CONTIN		TNPUTS	139
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### ### ##############################	#9\$ WATTE (06, 2002) ***********************************		INPUT	141
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NPUT NPUT	## ## ## ## ## ## ## ## ## ## ## ## ##		INPUTS	145
## ## ## ## ## ### ### ### ### ### ###	## 15 (4,201) xP.RP ## 15 (4,201) xP.RP ## 15 (2011) E. CONTINUE ## 2011 E. CONTINUE ## 10 (1) E. CONTINUE ## 2011 E. CONTINUE ## 2011 E. CONTINUE ## 2011 FOWM 1 (1) x + 10 H** PLUG DEF ## 2012 FOWM 1 (1) x + 10 H** ENGINE ## 2012 FOWM 1 (1) x + 10 H** ENGINE ## 2012 FOWM 1 (1) x + 10 H** ENGINE ## 2012 FOWM 1 (1) x + 20 H** ENGINE ## 2012 FOWH 1 (1) x + 20 H** ENGINE		# LOGNE	97
MRITE (6,2001) XP,RP	######################################		S LOUNE	147
## ## ## ## ## ## ## ## ## ## ## ## ##	\$82 CONTINUE \$8=5.486 \$82 CONTINUE MRITE(6,260) IF(NOFLOWECA.1) GO TO 808 AME(1)=AME		BLOGNE	e constant
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S	C GO TO 3030 3788 STAP C ZAB1 FORMAT(/1x ,18H** PLUG DEF 2565x,2FF0.4/) 2565x,2FF0.4/)	1 10001	INPUT	157
C CONTINUE INPUTS INPU	S CONTINUE C GO TO 3930 3788 STOP C 2881 FORMATI/IX .18H** PLUG DEF 2505x,2F20.4/) 2505 FORMATI/IX .28H** ENGINE		INPUTS	158
STATE STAT	200 TO 3030 3788 STAP C		INPUT	159
SABI STOP SABI STOP INPUTS	2002 FORMATITAL *18H** PLUG DEF 2515x,2F20.6/1) 2515x,2F20.6/1) 2002 FORMATITAL		INPUT	160
STRE STAP INPUTS INPUT	2002 FORMATITAL *18H** PLUG DEF 25(5x,2F20.6/)) 25(5x,2F20.6/))		INPUTS	191
2001 FORMATICIX ,184** PLUG DEFINITION/	2002 FORMATITAL SIGHAT PLUG DEF 2505x,2F20.6/12 2002 FORMATITA 2002 FORMATITA	•	INDIA	291
S881 FORMATILIX	2881 FORMATIVIX .18H** PLUG DEF 127x, 36HAXIAL 25(5x, 2F20.4/1) 2082 FORMATICIAL 2082 FORMATICIAL		INPULS	103
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335 L=L. IFINT,EG.0) NT=NA PLUSIGS CALL SETTAU(MT,DS,T,P1,P3,ALPHA(11,PA1,P81)				163
TFANT, EG. 67 MT=XA CALL SETTAU(NT, DS. 1, PL. PLAL, PAL, PAL, PRA	J. 1. 1. 1. 1. 1. 1. 1. 1. 1. 1. 1. 1. 1.			(69
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### CONTINUE ##	,	SF8G=SF2G+FG	PLUSICS	142
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THE CALLES RANGE AND MAYELENGTH PLUSIGS		GO TO 211	PLUSTGS	247
F(C)C, NE, C, N, NRIFE (6, 216) SUMEX		WRITE 16, 212) 272, SUMHP, SMAMP, SUMGS, SUMBG	PLUSIG\$	248
C IR, IK IMPLIES RANGE AND MAYELENGTH PRUSIGE		IF(XCC.NE.C.) WRITE(6,216) SUMEX	PLUS16\$	642
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PLUSIGE PLUS		IF(FILTEP. EQ.C) GO TO 401	PLUSIGE	260
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DO 430 I=4,NT PL(1)=PL(1)+NA) PL(2)(4) PL(3)(4) PL(4)(4)		これの大人によっている。	\$9ISNZ	272
PLUSIO PLII) = PLIIO A) PLUSIO B PLUSI		•	PLUS I G\$	27.1
P2(I) = P2(I+NA) P1(I) = P3(I+NA) P1(I) = P3(I+NA) DS(I+N		P1(1) =P1(1+NA)	PLUSIGS	274
P3(I) = P3(I+NA)		P2(I)=P2(I+NA)	PLUSIG	- 272
DS[I]=DS[I+NA] T(I)=T(I+NA)		P3(I)=P3(I+NA)	PLUSICA	576
T(I)=T(I+NA) CONTINUE CONTINUE FORHATISX,F10.2.2X,F10.3.56(2X,Z18.4).2X,F10.5) FORHATISX,F10.2.2X,F10.3.56(2X,Z18.4).2X,F10.5) FORHATISX,F10.2.2X,F10.3.56(2X,Z18.4).2X,F10.5) FORHATISX,F10.2.2X,F10.3.56(2X,Z18.4).5X,F10.5) ATY HATTSXSF *,9X,F00.3, ************************************		DS(T)=0S(T+NA)	100	112
CONTINUE FORMATISK, FIG. 2. 2x, Fig. 3. 6 (2x, Zig. 4), 2x, Fig. 5P FORMATISK, FIG. 2. 2x, Fig. 3. 6 (2x, Zig. 4), 2x, Fig. 5F FORMATISK, 5K, 5K, 5K, 5K, 5K, 5K, 5K, 5K, 5K, 5	264	T(I) = T(I+NA)	FORM	278
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Second S		I K+8 APPAYS FOR APPROPRIATE NUFRED &	KBCAL	96
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5 41 IF(N)FRED-LT.3303.360 TO 42 KBCAL FINDEX=44.+(INUFREO-3000.3780.)		INUFREQ-3088.1/18	KBCAL	54
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	PLADEFS	156
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	PLHOEFS	159
00x=10	PLHOEFS	160
0Ru=10r*JP0	PLMDEFS	161
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, D= (F 1) x	PLHOFFS	179

LS=LS+1 D01#JJ=LS+LF C0=C0+99(EK+ PSI=2SI+0PS C0J=0 C0P=0	J=2*X+JG-2 XX(J+1.50) AV(H=1. IF(A+H.ST. AMU=X0*(X(AMU=18S(AMU	159 CONTINUE C1=P/P58/2, Q=C1*x(2*2), C0=(2*P/RG, JN4=HA JN4=HA 176 CONTINUE TH=x(2*LS)// AMH=SOFT(x(1) IF(J5*GT*,1)	205	195 F F F F F F F F F	105 120 120 PSI=0 15 PSI=0 15 PSI=0 15 PSI=0 15 PSI=0 15 PSI=0 15 PSI=0 17 PSI=0 18 PSI=	175	216 IZ=H
LS=LS+1 001*JJ=LS+LF C0=C0+Q+(EK+.3333) PSI=D*(EK+.3333) CCP=0.	J=2*X+JG-2 XX(J+1*50)=ANH IF(A+1*57*.6) AHDH*1*6/(ANH*1*0) AHU=X0*(X(1*LS)+X(1*LF+1*)*AHCH/TM**2 AHU=185(AHU/CNST*	CONTINUE C1=P/P58/2, Q=C1*x(2*2)/X(1*2) C0=(A2*P/RCB)**2 JN4=HR LN4=HR CONTINUE CNTINUE TH=X(2*LS)/X(2*LF+1) AMH=SOPT(X(1*LS)**2/(X(2*LS)*A1)) IF(J3*GT.1) AMH=AMH*SQRT(A1/2403*)	CT=CT-77X (1,2) CT=CT-77X (1,2) CT=CT-77X (1,2)	IF(D=DA; E0; 0;) 60 TO 153 P=27:PDDX+x0 IF(P=PA)1:52:152:154 P=2PA P>2PA:0 P>2PA:0 P>2PA:0 P>2PA:0 P>2PA:0 P>3PA:0 P>3PA:0 P>3PA:0 P>3PA:0 P>3PA:0 P>3PA:0 P>3PA:0 P>3PA:0 P>3PA:0 P>3PA:0 P3:0 P3:0 P3:0 P3:0 P3:0 P3:0 P3:0 P3	LT.0.182=9LT.PR) GO TO	X(1,1)=AINT(GA)+PA/1080. X(2,1)=TA X(3,1)=XCC2A X(4,1)=	IZ=M2+QCx+5
		PLMOSES 203 PLMOSES 210 PLMOSES 211 PLMOSES 212 PLMOSES 213 PLMOSES 215 PLMOSES 216 PLMOSES 216	1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1	A 10 40 40 40 40 40 40 40 10 10 10 10 10 10 10 10 10 10 10 10 10		May the test of th	5

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	1541.[1.4.1NW) 60 10 191	PLYDER 231
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	IF(1355FSI-PSID)-DPS/21*(2-J6).6E.8) 60 TO 192	
; ;		PLADEFS 237
	COLENTOD	Ì
261	CONTINUE	
	C)O+6#6	i •••
	- 3	
	[12] 14 (5:2) / X (1:1)	,
	**************************************	PENDERS SES
	C7=DX*6KU*Y(1,1)*CO/(2,*(PSI*X(2,1))**2)	İ.,
	IFIC7, LE, C7M) GO T0193	.; •••
	•	
193	CONTINUE	
		1
190	Cx=3x+(x(I,LU)-x(I,LO))*CP(I)	PLMOEFS 259
		-
		PLHOTHS 265
	V(I+)=V(I+)+C2*(X(I+L0)+X(I+L0**	• •

		ļ
į	, J) +CJ/778.2/CP(100 - 100 -
	Y(2, 1) = Y(2, 1) + CZ* ((X(1, LU) = X(1, LU) = X(2, X)	
	ACCUTOS FOR TAICS FOR A MERCON CONTRACTOR AND THE C	 -
9	TF(15, 50, 2) 60 TO 179	PLMAEFS 271
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	LF=#R	
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•	CO 10 170	PLANE 277
	1.15.1 LAUCE	64
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	0CZ =CZAV/CZH	PLMDEFS 281
	9	
	21/11.	PLHOEFS 284
	60 10 123	
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		A17Kg 3HOA=pF8g 3p3hP5=pF8g 3p8HHACH NO=pF	2100 ECHAPITIONS// REPARENCE AND TO THE TOTAL TO	WALL CONTINUES DROWN AND THE CONTINUES OF THE CONTINUES O	305 LOS CONTINUE IFED-DX_LT.SAND.ANZ.LT.1.1 WRITEE6,22801PA,PB,AM2	300 250 MUNICHT 2011 60 TO 216 1517(1.HT.2).LT.0.11 60 TO 216 550 CONTINUE	James Hall	215 x(I,1)=x(I,3) x(1,2)=AIhT(x(1,2)=CU) x(2,2)=x(2,2)+CT G=ABS(x(2,4x)=TA=4,) IF(0.5) xR=xR+1	290	03 215 T=1.4 00213J=3,MP 210 X(I,J)=Y(I,J)
		PLHDEFS	PLHOEFS	PLHORES	7 P P F F F F F F F F F F F F F F F F F	PL KUMM	PLADERS	20000000000000000000000000000000000000	PL NO FE	PLMORTS

DAERLAY (XRMP, 4.1)			
SECONDARY FLIME		PLUME	
-	•	PLUAR	· (n
COMMON/LINK2/STAN(3)		SFO 4E	•
111		P.O. H.	
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		PLUME	11
JHHON/PLH/PLHG3 (431, RCN)	ST(501, MRD(58), PLM(5, 38, 50), MFLW, PA.P8,	PLUME	15
DPOXEKOLTALISHKE JSHKET		1000	13
NOTIFIED OF CONTRACTOR OF THE CONTRACTOR OF THE CONTRACTOR TO THE CONTRACTOR OF THE	11.051 12.14F12		12
_		PLUME	16
COMMON/FL/UA.TA.XCO2A.XH2OA.BX(53)		PLUME	17
		PLUHE	60 F
		PLUME	5.1
A.S. A.L. B.S. D		PLUMP.	20
883		PLUME	- 12
RPZ, S	SS. TANA,	PLUME	22
¥C.	XF, XO,	PLUME	23
•		PLUME	54
111037 . YEO2	1103) - XH20 (100) - U8 (100) - TS (100)	PLUME	25
101-101	PLM(261.1.1)	PLUME	92
CONTRACTOR CONTRACTOR	11, PLM (401,1,1)	PLUME	22
	RP. KOATA.	PLUYE	28
-	24.	PLUME	29
PA. PO.		PLUME	ao
AL AIND MIMON GRANDS AND AND AND AND AND AND AND AND AND AND	TS NOT USE	PLUME	31
TE XX20.TBT. P8.	RE ALSO	PLUME	32
PATRICE ONLY IF	-	PLUME	33
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XCO2 OWLY IS OMITTED, XCO2 & XH2	RE CALCULATERS	PLUME	35
VE TAKE A DR EOR ONLY IF DEFAULT	VALUES ARE NOT DESIRED.	PLUME	36
		PLUME	37
THITIALIZE		PLUME	38
		PLUME	65
. 00 110 IZ=1.50 .		PLUME	9
123=0.		E PLUME	61
STILLIER		PLUME	42
00 350 17#1#1800		PLUME	4.3
330 x(77,1)=6.		PLUME	.÷
TA:519.		PLUME	4.5
N85=20		PLUME	\$
		PLUME	67
KERRE		PLUME	.
KH2OB= 00033		PLUME	64
MESON - MESON		PLUME	20
		PLUME	51
TANKED. 0		PLUNE	25
		PLUME	53
00 311 E=1.100		PLUME	54
_		PLUME	55
SB1 XH20(I)=XH20A		PLUME	- S
RPN=5FH(2)		17.12	~~
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16=3 90 3 J#1+NEO K#KP+1+J-16	ì	60 TO 2	NS=NS-1	2 IF(hR*NS,LT,75) GO TO 6	NSH4	KOUNKO COMPLE AND E	ZOUCEZOS	IF(UA.LT.(.01*U5(1)))UA=.01*U8(1)	CP(4) =CP(4) +2.016/29.17	CP(3)=CP(3)+44.01/29.27	N=PL4GO(2A)	41=648+32+17*RG8	X PII X D	25:20	₹Z=₽₽	2X=0P	- F 4 .		- 1	- 1		DO 362 Jama, 188	MR#XR+1	製61 TST(J)=PLNGG(18)	U61J1=PLMGD(23)	00 351 J=12.50	POPPLECO(25)	15 THE TOTAL	CO SUC CHAPPE	CALL THRUST	P=92		-	HT CATCE TO BE KNOWN TO BE SERVED.	OTROOMS CENTED						PR(A) = 1.	PR(5)=1	PR(3)=1.	PR(2) = 5	P2(1) #1:	0.40 Hp. 74	
PLUME 115 PLUME 115	PLUME 111		İ					Ì				İ	į					į			97 1145 By					;		٠		İ	PLUME 75	. 74		PLU-E 72		7	1	PECKE	PLUMF RY	· (1		۰, ۵۰	İ	£1			•

	M=R=0+2+J		447
	SM.T=1.NS	SKOTE	6.1
;		PLUME	120
	=	PLUME	121
	Ξ	3 x 0 1 6	6.02
	XM201LK)=XH20(M)*(1,+4K)+4K*XH20(M+1)		161
~ 1	ARTHACAN / 20		125
	[(5=16+45+1	3kn1d	126
		PLUME	127
	THE FOLLOW FOLLS FOR PARTY.	3⊭n 1d	128
		PLUAE	129
:		5 K D 1 d	131
	B B X X X A D C C T B C L T C C C C C C C C C C C C C C C C C	SHOTE -	131
		PLUME	133
	4	PLUME	131
	AM2=5007(UB(11**2/A1/TS(11)	P L U 4	1. 1. 1. 1. 1. 1. 1. 1. 1. 1. 1. 1. 1. 1
	WEITE(6,1000)	PLUME	135
	7c=0.48	11 E O 1	1.55
	DX=(P3-92)/(NS*10.)	PLUME	137
	I=1.NP.NS	PLUME	e e
		PLUME	133
•	SM-DHOM2	M 10 16	147
		PLUA:	141
3203	416 11,21HF	PLUME	7.5
	(e,2150) P		7 P
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3300	E Tr. Tr. L	SWD 1d	146
(1	SENT O	147
2		PLUME	145
~	TAKE O	SHATO	149
	Manual Caraca Articles	PLUME	150
` .	MATERIAL SECTION IN	PLUME	151
	NS#18+21 GO TO 5	PLUME	152
		PLUME	151
3480	•	PLUME	154
	3+2/3/=CAN	A COM	155
	0R0=JR	PLUME	156
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į	The second secon	EMO Id	150
		SHOTO	161
	PSC 1 D m G	PLUME	162
		PLUME	163
	COLL FECTOR AND MEDING DESCRIPTION	PLUME	164
	「これでものできるのでは、これでは、これでは、これでは、これでは、これでは、これでは、これでは、これ	PLUME	165
	9 6	PLUME	165
		PLUME	167
	0.00 (1) /75(1)	PLUME	168
	2.10	PLUME	169
11	PS(J) =PS(J) +2*(3, *KN-OR)	34014	6.7
	_		171

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	CRHATCION GOHINCOMPLETE INPUT - CONCENTRATION ERROR. I		(FT/SEC) (DEG R)//) F12-2-3F12-6)	X , 1 9 T		CNST+CNST/2.0	CHST=(115,+GAMA+7,)/3,	.215	TF(64845,1530,500,500,500	CNST+194+	GAHRETA/(AHZTS(L))	Y(I,3)+0.	∀ (1,-2)=0.	X(1,1)*X(1,1)	9) 6) I=1,4	CONTINUE	X(4,20)HXT20A	x(2,40)=TA	1,4P) = AINT (UA) + PA/1000.	10 55 NACKO-4	1. A. T. T. T. T. T. T. T. T. T. T. T. T. T.		79764)#87+08#66#AL+61+(AL+1+1)	L. 11=6*XH2O(J)+61*XH2O(J-1)	X (U x M) HO (X C O C C C C C C C C C C C C C C C C C	X(1e 4) + 0 + 0 + 0 + 0 + 0 + 0 + 0 + 0 + 0 +	STATE OF	ī	113 DC 1 11-BC 1-11	601056		76[PS(J],GT,7*DPS/1000.) GO TO 48		IE:L.6E.981 GO TO 51	Z=PS(1)	-	1147
PLUME	BEO HE	אר סאניי אר סאניי	SHO 14	PLUKE	PLUME_	PLUME	ים ה הרטאה הרטאה	PLUXE	PLUME		PLUME	PLUME	10 1 2 C 3 C 31 C	7 C 3 T C 3	שר טאני שר טאני	3MD Td	PLUHE	בין ה היים היים	של טאנו הארם אנו	PLUHE	PECKE	PLUME	PERM	וואנו הארו הא	PH: JHC		PCJKE	TO THE	DEC.	0 TO TO TO TO TO TO TO TO TO TO TO TO TO	PLUHE .	PECHE		PLUHE:	PECED.		PLUKE
225	223	221 222	219 223	215	216 217	215		-: 212	211	515 502	203	207	200	204	2 C2	202			: 199 199	197	195	194	10 4	191	NO.	1 P P P P P P P P P P P P P P P P P P P	187	1480	∞ ∤	4 25 3	182	181	179	173	177	~ ~	

SUBROUTINE CHEM SUMPLE CHEMICAL ANALYSIS TO VIELD THE COMPOSITION AND BASIC AS ASSUMICE DEAL CCHRUSTICS OF THE CORE MOZIE INIT PLANE EXAUST AS ASSUMICE DEAL CCHRUSTICS. COMPANYELL DEAL CCHRUSTICS. COMPANYELL STATES C
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CHR	70 PC 40 PC	
XH20(J)=SPHV(2)	131 XH2013	h n
XCU2(J)=SPHY11) CHEM 64 CHEM 63	171 60 111 60	
The second secon	3x (2) 43	: !
O'Engl	GMR=1.7	60
CXCX	R68=154	

			,			205				100									9				6-				80			1	3.5		•	-					r A			! : 4		
60 TO 10 TAC=4447-40 50 TO 10	11 TAC=+44-Z	60 10 10		IF (H-22229.)	IF (H-16587.)115,	IF(H-15391,)	•	TE(H-93-6-)112-112-10	15(H=3256,0111.111.10	1510-4997	C ICAO - COLO DAY - HIL SIO ZIU		7 A=0	EC) 15-10-17		DECH STORES OF STATE STA	9		C 47 TO 52 KH OR 155,348 TO 172,011 FT.	6	TO TO ALL	0 TAC=411.57+.00504*4H-32300.1		C 32 TO 47 KE OP 1.5,518 TC 155,348 FT.	69 19 14	PAJ=,79426277E0=51216.65/1216.65+001+14-20000-1217-124-124-122-122-12-12-12-12-12-12-12-12-12-12-1	7 TAG		30		PACH 3, 2024 94 197 (10, 94 PAC)	5 TECH309*97		C 11 TO 20 MM OR 36,157 ' 65,823 FT		PAC=14.695449E0+4(230-\$57(200-150005-M+)**(-9-2020001000)	5 Tal=519,67-,3117FH		C D TO 11 KILOHETEKS OR O TO 36,152 FT.	17 17 47 . WALE 18 18 . I	3 IF (H-32)00,97,7,4	2 If (M-20103) 5.6.3	75 (Heiliggs, 5.5.2	こ 丁 一 八 一 一 一 一 一 一 一 一 一 一 一 一 一 一 一 一 一
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11	119		THRUST	111
115 102-131.7 104 105		TAC= 374.731602	THRUST	वर्ता । कर्मा ।
115 10 = 10 10 10 10 10 10 10		50 70 16	THRUST	113
11 11 12 13 14 14 15 15 14 15 15 15			THRUST	120
115 115	128		THRUST	171
11		TAD=334,74,000512+(H-18587	THRUST	225
17 14-21 17 14-22 17		50 73 10	THRUSH	521
17 10 10 10 10 10 10 10		TAL=165,7 301115541H-22229.1	THEOS	321
17 12 12 12 13 13 13 13 13		00 TO 10	THRUST	125
THRUST T		•	THRUST	126
THOUSE T		•	THRUST	127
11 16 17 17 17 17 17 17	*	NEC TON CASE	THRUST	123
			THRUST	129
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118 FC4.55.77.41123.4.4 119 CC. 11.2.1.4 119 CC. 11.2.1.4 120 CC. 12.2.4 121 CC. 12.2.4 121 CC. 12.2.4 122 CC. 12.2.4 123 CC. 12.2.4 124 CC. 12.2.4 125 CC. 12.2.4 126 CC. 12.2.4 127 CC. 12.2.4 128 CC. 12.2.4 129 CC. 12.2.4 120 CC. 12.2.4 121 CC. 12.2.4 121 CC. 12.2.4 122 CC. 12.2.4 123 CC. 12.2.4 124 CC. 12.2.4 125 CC. 12.2.4 126 CC. 12.2.4 127 CC. 12.2.4 128 CC. 12.2.4 129 CC. 12.2.4 129 CC. 12.2.4 120 CC. 12.2.4		7 60 7 7 4 7 6 7 7 7 7 7 7 7 7 7 7 7 7 7 7 7	TAU TA	
119 146=5627-31235599 119 146=41494811 120 146=41477-0035249-94-26240-9 120 146=41277-0035249-94-26240-9 120 146=41277-0035249-94-96240-9 120 146=41277-0035249-94-94-94-94-94-94-94-94-94-94-94-94-9		Ic(H-5:564) 116+3	TOTOTE	433
119 TAGETAL TO THE TOTAL TO THE TOTAL TOTA		TAC=562,7-,312345		***
119 TGC=L2-700352019*(H-11986.) 120 TGC=L2-700352019*(H-2026.) 121 TGC=L2-700352019*(H-2026.) 121 TGC=L2-700352019*(H-2026.) 121 TGC=L2-1-1-1-1-1-1-1-1-1-1-1-1-1-1-1-1-1-1-1		- 1	1200	131
120 To 10 10 10 10 10 10 10 10		TROK414.7+.033438	7072	ナリッ
123 14C=-42.7+.00352419*(H-2024.) C		GO TO 10	INCOSI	135
C C C C C C C C C C	118	3 TAC=-42.7+.03352819*(H-20286.	THRUST	135
C C C C C C C C C C	1		THRUST	137
THRUST T		HAT COLD HOT COLO HOT	THRUST	133
217 CDATEAN TO THOUSE THRUST T	•	19 TA × TAC	THRUST	139
THRUST T		7,	THPUST	140
THRUST T		,	THRUST	141
THE ATTAINS THE ATTAINS	•	4 4 4 4 6 7	THRUST	142
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## ## ## ## ## ## ## ## ## ## ## ## ##		WILL CO.	1000	
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RIMATE R			: 7	RIMAINS	96	<u> </u>
### ### ##############################				RIMENS	66	
######################################			(XM(IH. JA)	RIMAINE	103	
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JA=N3 [IM) +1		14	5 4 5 P.U.	RIMAINS	103	
### ### ##############################			14(14)	RIMAINS	104	
99 FORMITY STATE S			5.93) (XNITH-I), YN(IH-Y), S(IM-I), K5(IM-I)	MAINE DAKATAN	105	
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RIMAINS REAC(5,656 (LK(IM.I),IFI,JB) REALL			(MI)	RIMAINS	604	
			6.636)	RIMAINS	109	
KINAINS RIMAINS REALISTID AND REMAINS REMAINS				RIMAINS	113	
REALISTON REMIETS OF THE REMIETS OF THE WAY REMAINS REMAINS REMAINS FORMAT (3F10.5)	118		;	RIMAINS	111	
RFADICS + 1 MXT & R SUHK + HD RIHAINS FOR AT (3F10 - S) RIHAINS FOR AT (3F10 - S)			(5/1) NM	RIMAINS	113	
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	172	بد ا	CALL FIRSTLITEP(J), PRP)	Ş		
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	173	Z .	CALL COPLONING YOUR STREET COMMITTED STREET STREET	22		
	163		,			
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	167	- L	CALL CC _CKIPTP,PTS1,TTP,TTS1,GAMMAP,GAMMS1,RP,RS1,MP1,RS1,TARG,		165	
	165					
	164	1	MP1=4P/2			
	163	Ξ	RS1=RP			
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112	- '	RIMAINS	196
111	FORWAT (1H12//)	RIMAINS	
	MKALLI Contoba		
111	MRITE (6.108) XL(J), YL1(J), YL2(J), ACROSS(J), PT(J), XMST(J), TST(J), VS		
i i	(F) LENSICI)	RIMAINS	203
118	FORMATIC	RIMAINS	202
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.	4.1 (F1/3EU) (ES/	RIMAINS	213
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	IF(LINE-58) 211,211,212	RIMAINS	
7		RIMAINS	210
		RIMAINS	217
	The state of the s		016
211	WAITE (6,103) XL(3), YLE(3), YLE(3), MCROSS(3), FILS), MAITE (5,101,110), MAITE		228
	11[J]:0ENP[J]	PINETAS	221
1	1 NEW 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1	RIMAINS	222
612	F.7.	RIMAINE	223
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380	; ;	RIMAINS	525
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IF(KOOL(TAB)=2) 1012+1013+1024 CALL OVERLAY(GHXGHP+9+1+6HRECALL) CALL TPANCL	֓֞֟֝֟֓֓֓֓֓֟֝֓֓֓֓֓֓֓֓֓֓֓֓֓֓֓֓֓֓֓֓֓֓֓֓֓֓֓	コロニフィー・ロビュー もくびいと ヘイモー・アビア ジード かんけ エン・トメスコード はまり しびまつ	47 - 0	1	TOTAL TELEPROTATE SEED	OT INCHARDOR LOW	KI CONT I KKY CITY CONT	0 1360 JEK=1, IK	KOULSF=KCCL(TAB)	NOMOVERNO(IN)	NURO (E) NUR (IX)	MANUFACTURE (MISAH = 3 CORR)	NYOYETINYOVE+2	SHIFT (ISF) = TS (INHOVE)	99 1234 ISF=1+14	I AMOVE THE		LAUCH LSURF(J. III)	THE TAP 1	TO TO TOTAL OCTORS OF GO TO FORD		FIJSUSFIJ, EID . KE. JABON 60 TO 1815		IFIJSURFIJ, III .EQ, KSURF(IAB)) GO TO 1015	0 1011 TEDATE XXX	1117 112		C)207=0		NOSE NO. ""IZ"/25%, "SERPERMICKEN", F	T1,284, FEUID LUMP TEMPERATURES 1/7			MENUMENTAL AND AND AND AND AND AND AND AND AND AND	LINEFLELINEFL+3	HRITE(6, 461) NODER2(I, IN), IRIL		361 60 10 361	E(PRINTY _EQ0) GQ_TO_962	LTARL	63 T3 R55	DEPTH TERRITORY STEET BY SIGNAL STEET STORY	TF(K(T)-1) 451,852,853	# 1. XE	ENZERES.			ARC (LCC) II L+1
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		•	i : :						8	- A	25		1		y n			15							•		:	
											EN3	RETURN	PORTING REL	181 FORMATIO, 13x, 3, MAYERAGE WALL TEMPERATURE FOR MODE , 12, 14 IS	大切いのどものでのです はんのうこう		INCORPATS OF BY EXITE(6.10F) EOGENIES IN AT	HT=TADX/AT	SOUTHWISE CONTINUES OF THE PROPERTY OF THE PRO		ロロ・カン・スス・ス		KELYCII+K	00 110 1=1,4	KOL KOL	INTEGER PRINTS	COMMON/CC/ MODEWALLO],WILLION,KOOLT	SUBROUTINE AVERIA, THEIR, HODE M, THE PRIMIS
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	CINX OT DESCRIPTION OF THE PROPERTY OF THE PRO	FLOWYO	•
	51 T A	FLOWNO	•
		FLOWYO	*
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-		FLOWNO	10
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	1. (1.1) 6122	FLOWNO	13
•	() () () () () () () () () ()	FLOHYD	16
	2 0 00	FLOWND	15
7	6.6.2 (7-1)	FLOHVO	16
	(T) x x	FLOWNO	17
4		FLOWNO	
	15 (6.11)	FLONVO	. 61
23	KOCL . GT.	FLOWYO	20
	~	FLOWVO.	22
22	4 F (5x, 20H	FLONNO	22
	LCCATED AT .F5.1.1M., /)	FLORNO	24
		1 0 M & 0	200
	Matre (6.23) MLOC4.MLOC2	FLOWNO	
25 23	121	FLOWNO	26
	TEO AT "F5.1sth.s/)	FLOWNO	
!		FLOWNO	28
	=	FLOXVO	53
	,11,12	FLOWNO	er.
11	WRITE(6:1)	FLOHVO	, m
!		FLOWNO	32
	IF (KTOL +GT, 1) 60 TO 15	FLOHNO	P P
	MRITE(6:15) MLOS1; MLOC2	FLOWNO	*P
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51 67	TE(6,17) XLOCI,XLOCZ	FLOWNO	36
		FLOWNO	37
2	Ŀ.	FLOHYO	909
	CALED	FLOHNO	39
N ·	; •••	FLOWNO	0.9
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	I I	FLOWING	7.7
41	FORMAT (15212 BHFROM AXIAL LOCATION AFS.1,194 TO AXIAL LOCATION .FS.	FLOWNO	, m
	7	FLOWNO	177
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*	ENS	FLOHNO	4.6

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SOLA	DETINO	
POOL N 37	F 01-17-5-18-18-18-18-18-18-18-18-18-18-18-18-18-	35
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SOLV	59 × SONT ((8+8) - (1, *1+0)) / (2, *1)	
SOLY	ART-3/12.4A)	4
	O TO MONTH OF THE CASE OF THE	07
	##C}1-C12	77:
SOLA	C32=432702	-\alpha
	C31=A31/D1	25
50r.4	50,228=250	
20LN 205	C21=A217D1	
2014		• .
	1+2) ++ (+2(x) +42(x)) + (+2(x+2) + +2(x+1))	20
2014	22= (42 (K-2)+H2 (K-2))+(H2 (K-1)+H2 (K))+(H2 (K-1)+H2 (K-1))+(H2 (K)-H2 (K	
N 105)+(X)(X)=21(X))=(Y)(X+X)+Y)(X-X)	
502 4		
		15
205	32=(X2(A+2)+X0(A+2)) * ((X2(A+2)+X0(A)+(X0(A)+X0(A))) + * * * * * * * * * * * * * * * * *	
DLY 1	2x-111+(x1(x-1)+0(x-211)	
A 705	14-14-17-17-17-17-17-17-17-17-17-17-17-17-17-	
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S	A21=[H1(X-21)+[H1(X-2)]+(P(X-1)-P(X))+[H1(X-1)]+[H1(X-1)]+[H1(X-1)]+[P(X)-P(
PSOLN		
P 50	1 (7*1) X*2) X*2 X*2	
P SOL X	#11=P(X 2)=(#1(X-1)=#1(X))+P(X-1)=(#1(X)-#1(X-2))+P(X)=(#1(X-7)-#1	
SOL M		
SOLM	SUBTOUTIVE PSOCK(E1, E2, Ty ET, TT, TT, TT, TT, TT, TT, TT, TT, TT,	

	TE(X1, X2, V1, V2, X) Y11/(X2, X1) * (X-X1) * (
	RETURN STRATE & STRAT
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121	

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	34	TEANOL	FF=31FT((0,5+ZZ)*(0,5*ZZ)+1:)-0,5*ZZ	170
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	126	C	TE(H7(J)) 173-173-171	2
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	122	TRANCE	TABLEL (XT1, HGT,	
	121	TRANCL	33 J=1,L	120
	120	TRANCE	GO TO 13	4
	119	(3 (PT2E=PT1-((X12/SIGH12)+(H++N12))	
	Ę	7		:
	116	z.	206 CALL PSOLWINET, XMTT, XP, MT, DUM, 3)	115

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51		z.	
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5	FILMOL	TS,UA,K12,N1	
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38D SQNACH=[2,/(GAMA-1,1)*(127**(14.*GAMA)/GAMA1.4.1.1	ì		FILHGL	92
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29=(1.+(((GAMA-1.)/Zall'SQMACH)(RR*ZB))*ZA		T1 *CDX*144.01/SQPT(TC)	FILKCL	20
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302 WTT-AIT+W7(I) 202 WTT-AIT+W7(I) 202 CONTINUE 166 CH-KH-1 167 KW1-PT 168 KW1-PT 168 KW1-PT 168 KW1-PT 168 KW1-PT 168 KW1-PT 168 KW1-PT 168 KW1-PT 168 KW1-PT 168 KW1-PT 168 KW1-PT 168 KW1-W1-W1) 169 KW1-W1-W1) 169 KW1-W1-W1) 169 KW1-W1-W1) 169 KW1-W1-W1) 169 KW1-W1-W1) 169 KW1-W1-W1) 169 KW1-W1-W1) 169 KW1-W1-W1) 169 KW1-W1-W1) 169 KW1-W1-W1-W1) 169 KW1-W1-W1-W1) 169 KW1-W1-W1-W1-W1-W1-W1-W1-W1-W1-W1-W1-W1-W		H7[I]=SQST([32,174*GAHHA*SQHACH)/[RR*ZB])*ZA		T 6
282 CONTINUE 160 (WHENLE) 183,180,205 160 (WHENLE) 183,180,205 161 (WHENLE) 183,180,205 162 (WHENLE) 183,180,205 163 (WHENLE) 183,180,205 164 (WHENLE) 183,180,205 165 (WHENLE) 183,180,205 166 (WHENLE) 183,180,205 166 (WHENLE) 183,180,205 167 (WHENLE) 183,180,205 168	20	HTT=ATT+N7(I)	100	S -
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185 KW KW KW EW KW KW EW EW		4-31 1011181118	- UN - L 13	20
### ##################################	291	Test)	101111	9
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### IF (KKK) % % % % % % % % % % % % % % % % % %		11-4111	FILMOL	66
\$60	4 4 5	02-142-140B	FILMOL	100
64 WTHAX=(SIGHI2*[PII-SMALL)/KI2)**(I_0/Mi2) 61 WTI=MTMAX \$91.401.61 61 WTI=MTMAX \$91.401.61 62 WIT=MTMAX \$91.401.61 63 WIT=MTMAX \$91.401.61 63 WIT=MTMAX \$91.401.61 64 WK-PII- (KI2/SIGHI2)*(W*NI2)) FILMOL FILM	600	14 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1	FILMOL	101
	. 4	WITH 4 X = CSIGH12 + (PI1-SMALL) / K12) + # 410 / M12)	FILHEL	102
61 WTT=WTMAX A81 H=0.25*WTTF4,75*WT LOO F0 13 CO F0 13 CO F0 13 CO F0 13 CO F0 14 CO F0 15 CO F0 1		THE STATE SOME SOME SOME	FILHOL	103
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60 F0 13 F12 F12 F12 F12 F12 F12 F12 F12 F12 F12	;		FILMCL	106
#480 PCR==1.1 #480 PT=YP(KN)+PCR #INDL #IN		2	FILHOL	107
	684	_	FILMOL	108
	100	3	FILHOL	109
PCR-PCF/10.0 GO TO ADS GO TO ADS FILMOL BEAN WAXXXXI FILMOL	200	. P119 GO TO	FILHOL	110
GO TO NOS KKK-KKK41 N-ASTCHADO DETA-DE KKADO CL./NES		0.0	FILHCL	111
KKKHKK+1 Historyordianolandiakeliandiakeliandia	•		FILMOL	112
**************************************	489		FILMOL	117
		N=010110110+0110+0110+0110+0110+0110+011	FILMOL	117

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DO SJO Istank	TTMEFF (I+H)+TMEFF (I+H+M))/2-8)	DXA(L)=PIF+DX(I)+(R(M)+R(M+1))	00 14 11140		TF(K-I) 11,16,11		0 f ep = f f f f f = f = f = f = f = f = f =	TYPE TO 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1	71)	THEFE ((TO-TROG) /(1,0+3,6=(PAR[3=KXX=HG)))+TREG	TRANSCOTIC *** C+ 0000***************************	GO TO 60%	THEF=TRGG	TF(WZ(T)=0.0) 602,602,693	KKKH*KKK+Y* = 10 = 30007114K)	xxxx(IpJ)	IF(JA-1) 21,26,27		00 TO 25	CALL FACILITIES AFKOS XXSTXGS COMS KRI	1-1-1-1 On Va II	CALL TABLELEXE, RHG, XK, AG, DUM, KR)	XX+XC(X)+1X-(X)+COSTH(X)	1801.1811.1911.871.121.	JA = 2	60 10 22		00 11 x=14x	-	FV= (42/48) 4*(1,5*((46/48)+1,84)	, ;; •		CALL TABLEL (X1, XVG, XLCII), VG, DUM, XX	たいはアイディストの「マイスのイントです」(148)。 アプラスファンド	IECKS(I) *1E* 0*11 GO TO 601	10	20 3 THILE TO THE TOTAL TH		; <u>;;</u>	PT=PT1-((K12/5:GM12)+(H**M12))	CALL PSOLMAXIT, XXIII, XP, XII, 007, 32	II-HII 13,13,106
FILHOL	FILMOL	FILHCL	FILHOR -	FILMOL	FILHCL	ILHO	FILHCL	FILHOL	IL KO	11.40	110		FILHOL	FILHOL	FICHOL	4 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1	FILHCL	FILMOL	FILHCL	FILMOL)	FICHOL	FILHOL	FICHOL	# T	FILHOL	FILHCL	FILESE	FILMS	FILHOL	E 1 C I C C	12.5	11	יין ראטר יין ראטר		17.5	TICHOL .	# 10 E 13	FILEGE	ILHO	FILACE.	EILHCL
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~		FILHCL	571
L KL X (I)	KLA4ID=LX4IM=ID=B	FILMEL	174
SEE CONTINUE		FILMCL	175
		FILMEL	176
	~	FILMCL	111
25C2= (R	63	FILMSL	176
N-25 125 100	- 1	FILMCL	179
PRAT: (P	PRATE (PS(I)/PI) **SEC1	FILMOL	100
F0405=F	ZK J	FILMSL	181
TAN CONTINUE		FILMCL	182
•	5 .GT. #1	FILMEL	183
DESKACE FRA	1./.28x.12MFILM COOLING*/.Z/x.1xalchsukrauk	FILMEL	184
	7	FTI MEL	105
	1 MATTERS 153)		186
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	24.04.0	TF((TABS(J))+10) 5.5.67			IF(K-1) 5.	1F(XT+4) 5,54,64		j±	IF ([[NEW]]) -1)		PT2E1(NT)=PT2E			. 🛏	CONTINUE			515423=		IF(PT)	00 50 I=1,t	-60 TO 110		8	78: 11.614733-1:1280714446070071617	i S	52 WZ(I) #5+0	TF(77-1,) 101,102,102		JF(ILK-			7	7 .	(Tab)		TOTAL TOTAL STATE OF THE PROPERTY OF THE PROPE	DATA TO THE TAXABLE DATA TO THE DATA TO THE TAXABLE DATA TO THE TAXABLE DATA TO THE TAXABLE DATA TO THE TAXABLE DATA TO THE TAXABLE DATA TO THE TAXABLE DATA TO THE TAXABLE DATA TO THE TAXABLE DATA TO THE TAXABLE DATA TO THE TAXABLE DATA TO THE TAXABLE DATABLE DA	SUBCOUTINE SETFLOILIKALAPSIMINAMTAUA, CPC. TTL. TS. PTZE. TIZE
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	CONFLH	STOP
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İ	CONFLA	TELEVICE -
	CONFIN	CKH(ASHXAT/M3
-	COMPLA	95 43=(S12*(PT1-PT2(J))/K12)**(1,/K12)
	CONFLX	\$12=35,31*(PT1+PT2(3))/(TT1+TT2)
	CONFLH	TTZ=TS-(TS-TT1)/[XP(UAS/(864.PM1))
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209 10 00 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0	6(1)/755(1) (TT2)/(LHX(I)*AR*PT2(J)) (TT2)/(LHX(I)*AR*PT2(J)) (TT2)/(LHX(I)*AR*PT2(J)) (TT2)/(LHX(I)*AR*PT2(J)) (TT2)/(LHX(I)*AR*PT2(J)) (MCF1, MA1,EF4,MA1,DUM,NN3) 2*HN**2) 8*DC.**HD*MCC/(LHX(I)*,DGG5336*(TT7*TR)**7041) 8*DC.**HD*MCC/(LHX(I)*,DGG5336*(TT7*TR)**7041) 8*DC.**HD*MCC/(LHX(I)*BA1) 8*DC.**HD*MCC/(LHX(I)*BA1)**2) [SEN_1F1]/(LHXF*K5(IN-I)*SF) (SEN_1F1)/(LHXF*K5(IN-I)*SF) (SOC)/WC(I) 8*DOS	CONFLA CONFLA CONFLA CONFLA CONFLA CONFLA CONFLA CONFLA CONFLA CONFLA CONFLA CONFLA CONFLA CONFLA CONFLA CONFLA	123 123 123 123 123 123 123 133 133 133	
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	13 * dag5336 * (TT?* TR) ** , 7841) } (HH1) 11 * Lag5336 * (TT?* TR) ** , 7841) } 21 * HN1) 21 * HOPPIZ(J) * PR* (AHX(I) * AR) ** 2) XF * K5 (IH * I) / S5) }	CONFLA CONFLA CONFLA CONFLA CONFLA CONFLA CONFLA CONFLA	122 123 133 133 135 135 135 135 135 135 135 13	
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IFF(K-2) 214,214,215 CK*(4-CT(J)-MT(J)) MCT(J) CK*(4-CT(J)-MT(J)) M25,2 IFF(M-1) 220,220,221 IFF(K-1) 220,220,221 IFF(K-1) 220,220,221 IFF(M-1) 220,220,221 IFF(M-1) - MT(J) - MT(J) - MAXI 224,22 GO TO 224 H9=(MT(J)+MGT(J)) / 24,22 GO TO 235		CONFLA		
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F(A351CK1-0.003) 215.8 F(MSEJ_AGIA 0.0) 60 10 F(K-1) 220,220,221 K-2 F(KCT(J)-WT(J)) 225,2 GO TO 224 GO TO 224 H9= (WT(J)-WMAX) 224,22 H9= (WT(J)+WGT(J))/2,0 GO TO 235		CONFLA	163	
IF (K-1) 220,220,221 IF (K-1) 220,220,221 IF (MCT(J)-WT(J)) 223,2 LO=7 GO TO 224 GO TO 224 IF (MCT(J)-WMAX) 224,22 H9= (WT(J)+WGT(J))/2, GO TO 235		MILL W	# L	
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K=2 IF (WCT(J)= LO=2 GO TO 224 LO=4 WG=(MCT(J)= WG=(MCT(J)= WG=0 TO 235		CONFLE	0 1	
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IF (MCT(J)-WMAX) M9=(MT(J)+WGT(J) GO TO 235		CONFLY	121	
MG=(4T(J)+WGT(J) GO TO 235		CONFLA	152	
60 To 235		CONFLE	- 155	
		CONFLA	154	
		CONFLE	155	
		CONFLA	156	
		CONFLH	157	
i		CONFLH	158	
241=ff 822 00 122		CONFLH	159	
		CONFLA	. 160	
71.4=(ff121.4		CONFLH	161	
17+CC123=(CC128 &22		CONFLH	162	
~- ~-7		CONFLA	163	
		M LUNCO	164	
PT2(J)=P	/4.	E LANCE	165	
60 10 201			200	
2 (1-1) + 2			100	
IFIPTZ(J) .LT, PI11 60 TO 231				
1		COMPLE	100	
10 T 2 C T 01	*	CONFLH		
IF (LO-1) 229.2		CONFLA	170	
TRIBUTE (T) - KT (C)		COMPLE	1,1	
232 JF(LL-1) 234,234,233	•	במאגרש	711	

• ,	7.3.4	225					* * * * * * * * * * * * * * * * * * * *	. 220			;			215					1	512					502	1	:			1	200	Zt	7 T	-∀		205			• .		192		;								•	1 :				10.11	17 n			
252 5		253 L	261 I	245	271) r	7		270 H	7		3 1	217 4	243 I	ຄ	1			ລ			265 P		797	;	2 4		2 F 7	1	244	•		219 1	ດ	P	-	×						~	_	1 /12			1	c •		771	60	ĸ			933	st, i	1	_	2
F(L)		1=2	12	v	1	TECHULAT 270, 252, 252	v	GO TO 246		Z=u_05	1137		-45711 219	LO-1) 247,247,248	GO TO 246	==0.649		7-4		PT2(J)=PT2(J-11+Z	WT(J)=XSET	T2 [1-1]=P12(1)		+42=1 002	50 F3 C07			TE(J. E0.4) GD TO 257	3 269		IF(WOT(J)-WSST) 246,266,265		TF(X-1) 242,242,243	•	7 T	TTZ=TS-(TS-TT1)/6xP/UAS/(864-9)/SET)	KT (J) HESCT	ğ	2	KOT I	2.35		H + Do ph	CL*2	TOURCHIST.	-11 63636		3-31-40 0-43-3-6		7+10 d	,	0 10 254	K z T		ALL PROCESSES ATTSO, PSO, PG. P. 3	PS_(JA]=PTZ(JA)	KTTS375ATHCTC1AT	J.H. (YF) 05	933 JA=3,3	IF(J-3) 231,236,236
こうだった よっとった とっとった エ	021	0.41	2 2	2 2 1	047	ONFL	ONFIL	2	oic Cla	2 :	225	27.	CNY	27		2	277	ONFC	2	2) () () ()	2 .	2	יינאס		SEF	SEL	ONFL	SE	2		27	יי קר	אַר	ONF	3	2		2 2	2 C	֓֞֜֜֜֜֜֜֜֜֓֜֓֓֓֓֓֜֜֜֜֓֓֓֓֓֓֓֓֜֜֜֓֓֓֓֓֓֓֓) (C	2	2 C	2	2 7	SET I	CAFE		ONF.	OKEL	ONFL	OVE	CNFL	OVE	O KE	CKEC	200	ONFL	2
229	227	922		225	224	223	. 222	500	224	220	219	218	717		200	215	214	213	545	31.3	914	21.	202	205	207	206	205	204	201	646	202		201	# 45 T	197	146	143	474		401	197	400	401	1 2 2		187	186	185	184	Last Last	102	161	160	179	173	177	175	175	174	173

236		CONFLM	231
1	TA.(I) *HCT	CONFLM	232
	A(1)=P72(CONFLA	213
	200 2836 Talleton	CONFLH	the second
235 25	P12(1):4(1)	CONFLE	216
CONTRACTOR OF THE CONTRACTOR O		CONFLE	200
	()) ±P	CONFLA	238
: : : : : : : : : : : : : : : : : : : :		CONFLH	239
. 616		CONFLH	26.0
13	THE TYPE TO SECTION OF THE PERSON OF THE PER	CONTLA	142
-	7 (CONFLA	
	231 Tal.		9.4
28	TO(1-1) = TRO(1-1)		
245 282	CONTINUE	CONFL	246
	HYSER	CONFLH	242
	1.NC	CONFLH	248
		CONFLH	642
	2814111 ap2 on	CONFLA	250
L 5	10 L1 03	L L L L L L L L L L L L L L L L L L L	
295	*	CONFLH	253
	_	CONFLM	254
: : : : : : : : : : : : : : : : : : : :	Q + XXX	CONFLA	255
555 C +	T TOPANCE T	CONFLH	256
13	[F (NTC+4) 2-3-3-3	CONFLH	257
,	TO(J-1) = TOS	M I MANO C	0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0
	-	CONFLE	263
264		CONFLA	261
•	SKrd. 0	CORFLA	. 292
	DATE A CONTRACTOR LITTLE BOARD TO CONTRACTOR OF THE CONTRACTOR OF	CONSTR	263
	SERVICE CONTRACTOR OF THE CONT	T LUNCO	264 265
592		CONFLA	266
	IF(PEN.LT.RENL(1)) REN=RENL(1)	CONFLH	267
	TABLE	CONFLH	268
	1 (TC(J,I-1), PR)	CONFLA	- 269
***	* (CONFLH	273
678		CONFLA	
	E-CA-USELS/K-Z4-SEKSFKSFKSF FFKJ.GT.11 50 10 10		272
	-	CONFLH	276
	TFB = (TF(),	CONFLH	275
275	+	CONFLH	276
	THILL TO THE UNITED TO THE THE PRESENT STRANGED STOUGHTS (J. INC. ASLESSICE)	CONFLE	277
		CONFLE	278
	:	CONFIN	280
7.00	1F(NO, GT, J) GC TO B	CONFLH	70 00
	HC1 66 T	CONFIN	707
		CONFLA	; PH
	ACC= 7831A	CONFLH	592
	•	COMFLH	285
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	XXX)/"11/"X		***
	=	CONFLH	345
348	TEB - (TFB-TC(J,I-1)) *EXP(C)	CONFLA	346
		CONFLA	347
	S = TC(J+TMC)-TCS	CONFLH	348
	¥	CONFLH	349
	IF (185(0165) -1.)	CONFLH	350
	= (11()) = -	CONFLH	351
	33 37 (3 C	CONFLH	352
	\ \ \ \ \ \ \ \ \ \ \ \ \ \ \ \ \ \ \	CONFLH	351
	05 m41 1	CONFLA	
100		רושונים	222
607		CONFLA	no t
		E CALLED	257
		CONFLA	100 m
	<u>"</u>	CONFLH	359
•		CONFLH	360
362	E 177 E 17 E 1	COMELM	361
		CONFLM	362
!!!	-	CONFLH	363
	SXX + DS[[H]	CONFLH	364
	EFF = 14/(14+3.6*S	CONFLA	365
202	CTMOTHER WARRINGS - TYPESON - TRANSPORT FOR	CONFLA	365
	i	NATION OF THE PERSON OF THE PE	36/
		CONFLA	100 P
			369
4!			700
	46.00		17.2
	C=1X1 E62	CONFIR	3.2
	1	CONFLH	376
	294 IF:M1-1) 273,273,276	CONFLH	375
375	00 272 I=JK1+JK2	CONFLH	376
		CONFLH	377
	272 CONTINUE	CONFLH	378
	23 12 Cd 3 Cd 3 Cd 3 Cd 3 Cd 3 Cd 3 Cd 3 Cd	CONFLH	379
***	ALERIAL TABLE	CONFLH	D (0)
·			25.1
	486 "646" 546" 541 JI	E LUNCO	U 40 7
	GO TO 275	TONE I	484
	<u>u</u>	CONFLE	* un
385	STOP	CONFLH	386
	IFINTC-41 35,13,	CONFLH	387
	. N.	CONFLA	368
	THIN, 1) = TW (N-1, INC)	CONFLH	389
i	00 37 K=2, INC	CONFLH	390
398	Z	CONFLH	391
		E LUNCU	269
	O		260

305	7 4 5 1 1 1 1 1 5 1 1 5 1 5 1 5 1 5 1 5 1		257
	K H INC + 1	CONFLH	397
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	AND THE PARTY OF T	HOTOT	20
•	10.10	HOTPT	
75	HUMAN VIA IN THE ENGINEER PROPERTY OF CAR CAPE AND PROPERTY OFFI	Tator	22
33	ALT AND THE THE THE SAME AND THE TRANSPORTED TO THE TRANSPORT OF THE TRANS	Talca	; ;
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	PKNCQF "NOOEN "NYN XXXXXXXXXXXXXXXXXXXXXXXXXXXXXXXXX	TALDN	ED (1
		HOLF	- A.
35	, 363	HOTPI	
35	(xI(I), xF(I),TTP,FTP,THEIA(I,II)	HOTPI	41
	-	HOTPT	. 4.2
	TCSURF. ICS. PRINT2. II. CF111	HOTPT	4.3
	THE CAS	HOTPT	*
	NKN. NECON.	HOTPI	4.5
	D EQ.	HOTPT	94
	955 CM	HOTPT	47
	A KNUM TO DOMEST X X X X X X X X X X X X X X X X X X X	HOTPI	9
	CT CT	HOTPT	49
141		HOTPT	50
,	TRIKK(1) .60. U. GO TO 36	HOTPT	51
		HOTPY	52
	THE THE THE THE THE THE THE THE THE THE	HOTPT	53
	0 - 0 - 0 - 0 - 0 - 0 - 0 - 0 - 0 - 0 -	HOTPT	25
ř	COCKACT TANGETTO AND TANGETTO A	HOTPT	. 50
20	A CALL CALL STATE THE TAXABLE SCHOOL STATE TO SELECT STATE S	HOTPT	55
	NT2.TILEETI	HOTPT	57

AVERHT 2	AVERAT	AVERHT 5	AVERHT 6			AVERNT	AVERAT	AVERAT 13		AVERAL 15		AVERHT 18	AVER41 20		_	.		AVERAT 25	AVERHT 28		 - -		A UPDUT	AVERNI SA Averni usi	İ	AVERHT	« ·	AVENUE AND AND AND AND AND AND AND AND AND AND		AVERHT \$2	AVERHT 63	*	AVERAL 45			-	AVERHT				AVERHT 55		AVERHT 57	
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- 1	TELEVISION FOLS. FOLS. 2050	VIEHS	613
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	C. TO 409	S A S E A S	181
•	502T (RH02)	VIEWS	207
	2	2 X 3 Z A	154
10	TEST (SETE CALIFORNIA DE CALIF	VIEWS	185
3		VIEW	186
_	TH(SP2P1) 901,903,902	VIEWS	187
-	#11 + 16248180Ca24+16 # 15	VIEWS	168
	TF (CST) 7040.7040.7060	VIENS	189
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C.	TF (AP2P1 -8P2P1) 9010,9010,9012	VIEKS	204
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1462	Y1, 51, Y2, R2	VIENS	328
		NIEM	329
1435 H	E(6,1467)	VIEWS	330
		VIEWS	331
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